

# AVIATION

*The Oldest American Aeronautical Magazine*



PRELIMINARY REPORTS FROM *Cleveland*

THE *S. A. E.* AERONAUTIC MEETING

*Engines* AT THE *Olympia Show*

*Supplement Specification Table of American Commercial Airplanes With This Issue*



*An important room of the new B/J Plant, Baltimore, Md.*

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**M**AN power is the secret of salespower for the B/J distributor and dealer. Back of the B/J plane is the weight of a great staff of foremost aircraft engineers. They were gathered together for the pioneering of new designs for military planes. This same original designing is carried into the B/J commercial planes, truly destined to be pacesetters of the air. Every B/J distributor and dealer knows that back of the plane he sells is an engineering staff that is making history for the planes they build, and permanent profits for the dealer. Some distributor territories are open.

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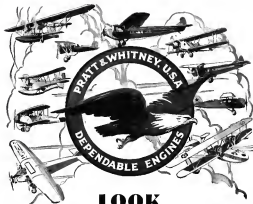
The Tarvia field man will give you complete details. Write, phone or wire our nearest branch office.

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Experienced pilots will tell you this trademark represents millions of safe, comfortable and swift miles through the skyways of the world with passengers, mail and merchandise.

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Break Limit 837,750 lbs.  
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CONSOLIDATED AIRCRAFT TRAINING PLANE EQUIPPED WITH AIRWHEELS FOR TEST PURPOSES

# AIRWHEEL

## a totally new airplane tire

Here is a big, soft, air-filled rubber cushion—a literal "wheel of air".

It operates at pressures as low as five pounds, ranging up to twenty—and it pillows a take-off or landing as no tire has ever done before.

Since the beginning of flying, airplanes have operated on tires adapted from the automobile—an entirely different type of service. Now for the first time the airplane has a tire of its own—conceived and built for the peculiar duties of cushioning several thousand pounds of weight as it leaves or meets the ground at express train speed.

On the take-off with Airwheels there is no "hopping". They provide continuous ground contact until the ship gets into the air.

From the first landing impact they keep the plane in gentle contact with the ground—with none of the violent rebound which an imperfect landing might otherwise cause, and with smoother, quicker brake operation resulting in shorter stops.

They land safely on snow, soft, wet

fields or sand—where former tires might have caused a crack-up.

They make it practically impossible to drag a wing in a ground loop—even if you tried to do it.

All the usual wheel failures are eliminated for the simple reason that there are no wheels. The Airwheel and its tube mount directly on the hub.

Landings have been tried with one tire deflated, and the loss of radial distance proved almost negligible.

Tests indicate that Airwheels should cause less air resistance than standard wheels used in flying—and in installed weight they are the same or less.

There is reason to believe that Airwheels will render any other shock absorber needless on ships of every type.

The New Goodyear Airwheel is available in a limited number of sizes and in limited quantities at the present time. For information or engineering assistance in equipping your future ships, write Aeronautics Department, Goodyear, Akron, Ohio, or Los Angeles, California.

# GOODYEAR

*Everything in rubber for the airplane*



420 hours  
21 minutes  
30 seconds

IN THE AIR ON

## SRB BALL BEARINGS!

A CURTISS-ROBIN plane powered by a Curtiss Challenger Engine has established this New World's Record for 'Refueled' Endurance Flight—a remarkable demonstration of aeronautical engine reliability.



The SRB Ball Bearings with which this engine was completely equipped—1 in the propeller thrust position, 2 on the crankshaft, 24 on the rocker arms—functioned perfectly throughout this grueling test of continuous performance.

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# DAYTON BEAR [Four-in-Line]

*Air-Cooled*

*Super Performance*

*Low Upkeep*

**Manufacturer's  
Rating . . .**

**110 Horsepower at 1550  
revolutions per minute**

**120 Horsepower at 1850  
revolutions per minute**

*Immediate deliveries*



**The Dayton Airplane  
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## National Headquarters for Aviation Supplies . . .

**A** WAREHOUSE for the industry—so is the Robinson Supply Division of the Universal Aviation Corporation considered by those who have had occasion to draw upon this vast supply depot for various aeronautical needs. Practically anything and everything pertaining to aviation is carried in stock, ready for immediate shipment.

Bells, nuts, washers, fabric, dope, engines and engine parts, and other supplies for manufacturers and individuals, as well as all manner of flying equipment, can be found in the store rooms of this supply depot. Tremendous purchasing power has made it possible to secure this material at lower possible

prices. Often the user of aircraft supplies can purchase items Universal at prices far less than if dealing direct with manufacturers. . . . Delivery is prompt. Every item carried meets A.N. specifications or has been accepted as standard through general commercial usage. Every item is guaranteed to give full satisfaction or purchase price is refunded.



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PERFORMANCE reflects the value of propellers. Since Standard Steel Propellers were first developed ten years ago, they have constantly won the favor of pilots everywhere, for unfailing performance under every flying condition.

Select Standard Steel Propellers for your next propeller replacement—adopt them as standard equipment! Furnished for engines rated at 50 to 600 horsepower.

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## Also used on these RECORD-BREAKING FLIGHTS

Captain Harbo used to coast eight in a Lockheed Vega  
The endurance flight of the Curtiss "St. Louis Robin" by Dale Gribble and Forrest O'Brien

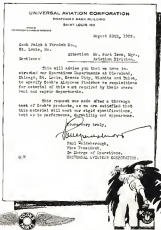
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We have an interesting story for any aviation man interested in the most durable fabric finish for airplanes. It is fully explained in the Cook Finishing Plan, which we will be glad to send you on request. Write for it today.



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minutes after it  
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The order is passed  
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Write, wire or phone your order and see what a pleasure it is to deal with a house that gives you the kind of service that you have every right to expect.



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color in the stock  
room, where it is  
promptly filled.

The shipping  
department is  
located in the  
warehouse.



Within a hour  
of its receipt, the  
order is ready to  
be delivered to  
the buyer's freight  
house.

Through the various  
departments, the  
order is passed  
and is finally packed and  
shipped.



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**B**ACKED by years of experience in the various fields of commercial illumination, Westinghouse has developed airport lighting equipment which is effective, flexible, moderate in cost, and which fully meets the requirements of the Department of Commerce for the highest rating. Westinghouse has lighted many airports throughout the country, one of the outstanding installations being the Newark Metropolitan Airport, which has been declared by experts to be the best illuminated field in the country.

### Lighting Equipment

Chromalite Landing Field Floodlights  
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The Vernal Amphiphan Air, just as shown in the Westinghouse Lighting Equipment.

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## AIRCRAFT Accessories - -

**W**ESTINGHOUSE manufactures such aircraft accessories as Minolta propellers, control-wire pulleys, fair leads, tail skid wheels, and plate for the interior fields of cabins. Minolta propellers are light in weight, free from vibration, extremely durable, and are unaffected by moisture and change in temperature. They have figured in such



The drag motorless Quarter Shaft, showing the main wing engine equipped with Minolta Propeller immediately after completing post-landing action on flight.

### Accessories

Minolta Propellers  
Minolta Control-wire Pulleys  
Minolta Fair Leads  
Minolta Hinges  
Minolta Tail Skid Wheels  
Minolta Plate for Cabins



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EAST PITTSBURGH PENNSYLVANIA

outstanding achievements as the crossing of the Pacific to Australia by the Southern Cross, the record-breaking endurance flight of the Question Mark, and the first flight to Hawaii by the Army tri-motored monoplane piloted by Lieutenant Mayland.

## THE LIFE PRESERVER OF THE AIR

(Over 25,000 Happy Landings)



We have the Irvin Air Chute School at Los Angeles after making their qualifying jumps in the States.

## The universal safety device . . . The IRVIN AIR CHUTE . . . is now available to all who fly

**B**RING the reputation equipment for all the air forces of the United States—army, navy and air mail, adopted in such by 26 other governments all over the world, chosen by such eminent air leaders as Col. Lindbergh and many others—the Irvin Air Chute represents, today, the world's most universally employed aeronautic safety device.

Through inspectors of every small detail, infinite care and the finest material make the perfect functioning of every Irvin Air Chute. No production will ever be made of what this parachute will do. Past performances and present worldwide confidence

show any proofs of perfect safety in the record of past service to civilian and naval aviators.

Two hundred and thirty-two emergency jumps have been made since 1918. All but one of them were made with Irvin parachutes. On more than 25,000 occasions "live" and emergency jumps with the "Life Preserver of the Air" have proven undisturbedly successful.

If you are interested in purchasing an Irvin Air Chute, you can secure it from the Robertson Aircraft Division of the Universal Aviation Corporation.

Irvin Chutes are manufactured by  
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\*This outline on the  
roof of a hangar shows a  
branch of Southern  
Aeromotive Service—the  
most complete in the South.



Airmen who know the South mark each of the 16 S.A.T.S. Points of Service as show maps

At 16 strategic points, S.A.T.S. offers complete general and passenger service, ports, area supplies, construction.

In the South, head your plane far as S.A.T.S. port, whether it is refueling, patching a wing, checking or rebuilding your motor or just for a pile of goggles. Service exists you there at the hands of Government licensed mechanics, prompted by an intelligent understanding of your problems.

Southern Aeromotive Service, Inc., is a division of Southern Air Transport System, operating air mail and air passenger lines throughout the South. It is in S.A.T.S. shops that ships of Southern Air Transport Systems are serviced.

The same service and equipment which maintains the splendid operations record of S.A.T.S. ships is open to you at any of the 16 strategic points of S.A.T.S. service.

No matter where you are, your call to the nearest S.A.T.S. Station will bring a plane with mechanic, supplies, equipment or parts to you immediately.

WATCH FOR THE S.A.T.S. CATALOGUE

**Southern Aeromotive Service, Inc.**

DIVISION OF SOUTHERN AIR TRANSPORT SYSTEM

### Air Force Pagesday

**D**URING the present week at Cleveland the industry is displaying its wares. The atmosphere is and it should be primarily commercial, but the military is not excluded. The "Three Sea-Hawks" who show at Los Angeles have been debanded, but their successors will be on hand to show that the Navy can fly as well from an island field as over the sea, and the Army's First Pursuit Group will exhibit with an increasing flourish.

Army and Navy pilots will be present. They will compete as individuals, as squadrons, and as groups, or wings. But they will have no chance to plan a military program. As displays of aerial acrobatics, their contributions will be surprising. As displays of the actual use of air power, they will be negligible fragments.

Of that, we have no criticism to offer. The attempt to combine a military pageant with a commercial show and a group of civilian air clubs would be out of place. The military element would weaken the appeal of the commercial features for which the race program primarily exists, and on account of which the industry supports them. Even at Los Angeles a year ago some of the civilian competitors protested that the Army and Navy had "ruined the show." Mr. McHenry, in his article in our last week's issue, pointed the lesson that the most brilliant of standing exhibitions may thoroughly abate the average spectator from any idea of flying on his own account. If there is to be a truly military display it ought not to be an appendage to the National Air Races. Whether the appendage merely took a secondary place, or whether it became the tail that wagged the dog, its presence would be objectionable. If there are to be real and complete military shows they ought to stand alone.

The Royal Air Force pageant, which has just been held for the fourth time near London, is an excellent, and in fact the one commanding, example of what such shows may become. Syndicates of pursuit, observation, and bombardment go through their exercises and routines in astonishingly accurate simulations of the actual

work of real war. Bombing attacks by wave-fitted parts built in paper-mache and plaster with the living case that Hollywood devotes to the construction of a "set." Planes engage each other in battle, and spin down in shattering realistic flames. Observations are made, and the pilots' reports are conveyed directly to the crowd by radio and loud speaker. The closest approach to the fading realism of that display that has ever been offered to a guest crowd in America was presented by the attack of a Navy light bombing squadron on a village erected in the middle of Mines Field at the National Races of 1928. Would it be worth while to emulate the British and run a separate display of Army and Navy air power under the conditions of field exercises? The suggestion that we should do so has often been made.

**T**HERE IS SOMETHING IN FAVOR of such a show, but after careful reflection we are against it. We do not believe that it could help commercial flying—and in the long run, nothing that gives a setback to commercial aviation or the aircraft industry or to the private use of planes can be in the interest of Army or Navy air strength. The duration of the services from their present training schedules and maneuvers, now often carried on in Southern Michigan in mid-water, on the plains of Texas, or far out at sea, in order to rehearse a program for an afternoon's entertainment, would be a serious sacrifice. We can see no probability of substantial gain except a political one—and even that might prove a two-edged sword. If Members of Parliament are like anything like Members of Congress in the mass, a visit to the R.A.F. Display is less likely to suggest to them the need for additional appropriations than to convince them that a force that can do such marvelous stunts must already be admirably equipped.

No people can fully interpret the psychology of another. Whether or not the R.A.F. show has been of real benefit, under the conditions existing in Great Britain, to the solid development of British aviation we do not know. Finally, we doubt if non-militarizing

that several members of this editorial staff have had the great privilege of attending the Dugby and have come away with an enormous admiration for the R.A.F. and all its qualities. In any case, American conditions are different, and we sincerely trust that neither the content of advertisements nor any other attempt to manipulate public payments on a grand scale here.

## //

### A Triumph of Faith

**D**UBUOH has been witnessing events unique in aeronautical history. Indeed, forward in the face of peril, of destruction, of industrial doubts and of financial struggles, the first all-metal airship has been completed. It has been filled with gas. It has flown. Leaving aside a few shabby experiments of the gas-Zeppelin era, it is the first of its kind. It makes its way from the hangar out into a world in which, only a few years ago, the very idea of a metallic balloon would have been the subject for ridicule as hearty as that which with a last-bearer of the seventeenth century would have greeted the notion of an iron ship.

Every innovation in the design of lighter-than-air craft has had behind it a group of men who figuratively walked up to the edge of a precipice and looked over, carrying their faith in their hands. Every one has a colossal gamble in it. There is no way of trying the thing out cheaply. Calculations can never be rigorously complete. The self-confidence that led Count Zeppelin, thirty years ago, to plunge on the construction of his first ship is fully now-inspiring. Only a little less enthusiasm and faith have been exacted of those who created the idea of the metal-airship and lived with it for seven years and built it and brought it finally to the stage of real flight.

In doing all this, they have afforded the chance for a picturesque demonstration by the industrial leaders of a great city of their faith in the future of aeronautics, that sort of faith which is the substance of things not known. Upon the board of directors of the company that built the metal airship there, on some twenty men. Perhaps not more than a sixth of them, and certainly not more than a third, would be able to say expert knowledge of airship design or operation. They are there because they believe in strength and they believe in progress, and they wanted to lead a band and to feel that they had had some personal share in an experiment in which there was nothing obviously unworkable, and which recommended itself to be a very boldness.

A single flight, or even a dozen, does not prove a vehicle of commerce or of war. There have been scraps of eminent standing, and there still are. It was only in the most energetic and persistent campaign and

against strong opposition, that a May contract for the present ship was secured. The all-entire construction is subject to certain obvious drawbacks and disadvantages, even as it has certain obvious merits. For ourselves, we reserve judgment on the ultimate possibilities of the type, but we are delighted that it has now given the opportunity of practical trial.

In the general enthusiasm over the success of a ship built as the result of a corporation, it should not be overlooked that the work was done by individuals, and that there were key men without whom it would have been difficult if not impossible to organize the project and carry it through. AVIATION's congratulations go with special warmth to Messrs. Claus and Fritzsche and Hill and Schlusser and their co-workers, to Messrs. Emmert and Ebers and the others who stood by with unswerving support when success seemed very far away.

## //

### The New Secretary

**W**HEN THE AIR COMMERCE ACT was approaching passage, three years ago last spring, there were accessible prophets of gloom who predicted that the Department of Commerce was to become the graveyard of aeronautics. No one, they charged, could meet the conflicting demands that the work of the Aeronautics Branch would impose. No one who was so foolish as to accept high place there, were informed, would be able at once to administer his office honestly and in accord with the letter of the law and to preserve his own sanity and the respect of his neighbors.

But they were all wrong. The Comstocks and the Lottens were side of the mark. The test of time and experience has proven that their world and luxurious surroundings was unneeded for. The Aeronautics Branch has striven through its stormy seas and emerged on an even keel in almost unbroken waters. In 1925 there were many builders and operators of aircraft who were opposed to any Federal regulations. A year later there were still a few who thought that the aeronautical work of the Department of Commerce would be commendable in preparation as it was unreasonably restrictive. If any such remains as the industry they are acting in the spirit of that day when the captain informed the first mate that he "waked" from him "looking far wiser, and damn little of dirt." There is, to be sure, no want of complaints against the details of Departmental action, but regulation is accepted as a fact, and it is recognized that it counts in the industry's own interest. To draw again upon a paraphrase of Voltaire's more-than-quoted saying, if these were the Department of Commerce we should find it necessary to create one.

We are slow in making to the point, but finally we

AVIATION  
August 31, 1929

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arrive. What is it that has wrought this change in feeling, and this increase in the habit of looking to the Aeronautics Branch for assistance and advice, regarding it as a resource rather than as a policeman? The natural pressure of events has had something to do with it, but it has been by no means an automatic process. It comes primarily as the consequence of a growing confidence in the Department's aeronautical personnel, and specifically in the men who have had charge of the work and shaped the policies. They have themselves to thank that this confidence has reached and that desperately difficult position has grown gradually secure.

Secretary McCracken having determined to retire, and remaining here in spite of all persuasion, the President has turned to the man who has been McCracken's right hand throughout these three years. Clarence M. Young is to become Assistant Secretary of Commerce for Aeronautics.

Clarence Young is so well known to the industry that it is hardly to try to say anything new either of the record of his work or of his personal character. Of the two key positions in the aeronautical work done by the Department of Commerce, his has been probably the more difficult, or at least the more harassing. He has had most direct contact with the detailed technical problems of the industry, and his office has been the first point of attack when the industry's personnel protested the government's rulings on interpretations and applications of the regulations. He has had to say "no," and say it often and positively. That he has been able to do that and still retain the respect and personal friendship of those whose requests he denied is the best possible personal tribute.

We shall make no effort to add the life by stringing together inebriate tones of praise. We participate in a general confidence that the work of aeronautical promotion and regulation under the Federal government will remain on the highest plane, conducted on lines at once far-sighted and sober, progressive and careful. From the day when he takes office, Secretary Young will be able to count on the cordial and whole-hearted co-operation of every respectable aeronautical interest.

## //

### A Well-Earned D.F.C.

**T**HE Distinguished Flying Cross was created, at the recommendation of the Morrow Board, to rank as a decoration second only to the Congressional Medal of Honor in significance. Among the officers and men of the Air Corps and of naval aviation it has that standing. It is not, and it ought never to be, given for personal or political reasons. It ought never to be awarded in response to mere public acclaim. It is now given, both in the War and Navy Departments, only upon the recommendation of boards of officers acting with every shred of

**T**HE Circulation Department desires to include the announcement that the classified and alphabetical index for Volume 26 of AVIATION is now ready for distribution to all those who desire same. This index covers the issues of AVIATION from January to June (1929) inclusive.

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available evidence before them. The identity of the members of the boards and the details of their recommendations are not made public. They are subject to no pressure, and that is it should be.

The War Department has just made a fresh group of awards. All are notable, but one in particular we single out as being a fitting recognition of work which had no spectacular element and received little or no attention from the press. Lieutenant Harry A. Sutton's research on opening, carried out at the greatest risk of his own life and after one of the Army's best test pilots had been killed in a flat spin on a machine of the same type that Lieutenant Sutton was studying, are worthy of terms of praise more eloquent than any that we can command. The officers of the Engineering Division of the Air Corps did not put the matter too strongly, in recommending Lieutenant Sutton for the award, in describing his studies "The finest demonstration of brain, courage, and daring which had been brought to their attention in many years." The War Department goes proud of both willingness and ability to risk the purely technical achievement contributing to the development of aerial navigation so highly as the purely military considerations.

If the spinning experiments had been merely military in their significance we should not be mentioning them here. They are more than that. The experience that they represent ought to enter somewhere, directly or indirectly, into the design of every airplane. No designer could possibly read Lieutenant Sutton's report without learning something new about how to guard against undesirable spinning characteristics in his own product. Much can be done in the wind tunnel to forestall flying characteristics and improve them, but the solutions of all such problems as that of the flat spin depend finally, after decades have been devoted and researches in the laboratory have been carried to the limit, upon trial in the air by a pilot. Test pilots, military and commercial, usually fail to receive the credit that is their due. The conferring of the D.F.C. upon Lieutenant Sutton adds fresh laurels to a already brilliant career.

# A STUDY OF THE *Engines*

## EXHIBITED AT

## Olympia

## THE

## Aero Show

By C. FAYETTE TAYLOR

Professor of Aeronautical Power-Plant Design,  
Massachusetts Institute of Technology

**A** STUDY of the engines at the Olympia Show lends credence to the general opinion that American engineers have exerted a profound influence on foreign airplane engine designs. What we are pleased to call the "American" type of air-cooled cylinder is now almost universally adopted, the last to come into line being the well-known British firm, which up to that time had consistently used an all-steel cylinder. One must admit, however, that the "American" air-cooled cylinder with screw-on aluminum head was originally a British development, carried on by the Royal Aircraft Establishment during and after

the War. That America adopted this construction almost unanimously is a tribute to the high quality of British research work, as well as to the efforts in America at S. D. Hearn. That America has improved on this work is not a bad thing to say, but we must acknowledge our indebtedness to Great Britain for the fundamental work on this type of cylinder construction.

More truly American is the inclined valve gear for air-cooled engines, which has been adopted by all the important foreign manufacturers of radial engines, including notably Bristol and Armstrong-Siddley, the two

*The American visitor at a European air-craft show naturally examines the exhibits with special reference to their similarities to or divergencies from American practice. It is from that point of view that Professor Taylor, formerly in charge of the power-plant laboratory at McCook Field and subsequently with the Wright Company, has written of the engines at Olympia.*

leading British manufacturers of this type. The radial-engine has design on two engines, the Farmas 280 hp seven-cylinder radial and the Fiat 100 hp seven-cylinder radial, in obviously taken directly from the latest type of American cylinder head.

Several variants of the N.A.C.A. cooling were in evidence at the show, ranging from the somewhat exaggerated form shown by Bristol, which subjected the engine in a case of such proportions that it might have been mistaken for a fully enclosed water-cooled engine, to the "Towson ring cooling" on some of the Armstrong-Siddley engines, which is little more than a vertical lift over the top of the front row of cylinders.

In the water-cooled field, many of the leading manufacturers have adopted the cast aluminum cylinder block construction first developed by Cummins. This includes such companies as Holtz-Beyer and Lorraine Dietrich, who heretofore have adhered strictly to the welded steel construction.

A very general use of propeller reduction gears was in evidence on both air-cooled and water-cooled engines,



Bristol "Mercury" and "Jagler" engines

and in this respect one is inclined to believe that the foreign practice is ahead of the American. Many engines also carried superchargers, mostly of the centrifugal type driven through gears and a clutch. Other characteristic features not generally seen heretofore were forged aluminum pistons and ignition systems shielded against radio interference.

The twelve-air-cooled engine has evidently come to stay, there being several examples of this type in the English exhibits and at least one each in the French, Italian and German sections. Dornier engines were most on several water-cooled engines of Continental origin. In that respect European practice is distinctly ahead of the American. Many of the engines at the Show are already well known to the American aeronauts public and require no comment beyond the mention of their names. However, many new models and much improvement on existing models were in evidence. A tabulation of the more important engines at the Show, giving their principal characteristics, is published with this article.

Perhaps the most interesting exhibit at the Show, at least from the point of view of new development was that of Bristol Aircraft Co. Ltd., which had a very impressive display near the main entrance of the hall.

The "Bristol Mercury" engine, comparable in the last Schneider Trophy Contest, was probably shown for the first time and attracted an enormous amount of interest. While it will be the curiosity of its design from the "Jagler" engine, almost all of the details have been changed, and one would say greatly improved. The most important change is the cylinder-head, which is a dovetail root-head after machined camshaft with cooling fins from a single distributor leading. The head is screwed on to the steel barrel against a solid copper gasket, and carries a very cleverly designed sub-gasket isolated in a cast aluminum case and operated by jurelled push rods, compensated for expansion.

The process of machining this cylinder head from a heavy block of aluminum was observed by the writer last year during a visit to the Bristol shops. That a four-



The engine booth. Left is a V-12 engine; in the center and to the right are two "Star" engines.

valve fully fitted head, with valves set at an angle, could be as designed as to be practicable for modifying from the solid, is certainly a credit to the engineers who conceived it. One can hardly imagine, on the other hand, such an expensive process being carried out in the United States, and we are inclined to believe that the American



method of developing a free foundry technique for casting such cylinder heads is, on the whole, a miser pakey.

The Mercury engine has a slightly shorter stroke than the Jupiter and carries a 9500 Parson reduction gear and a built-in centrifugal supercharger. The mounting flange, large enough to receive all the accessories, is made up of steel stampings, and the mounting bolts are carried in flexible rubber joints to absorb vibration. The carburetor is fitted with a barometric throttle stop to prevent over-supercharging at sea level. Another example of this engine was shown indeed in an N.A.C.A. coating of rather extreme form, which has already been mentioned.

The other British engines, namely the "Titan," "Neptune," and "Jupiter," which have been well known for some years, were also on display, but presented a new and attractive appearance with their aluminum cylinder



Aluminum cylinder alternative barrel-type engine

heads with inclined valve gear, similar to the cylinder heads used on the Mercury.

Another new model of considerable interest appearing at the Show was the Rolls-Royce Type HX 825 hp, 12-cylinder "Vee" engine, displayed for the first time. It follows quite closely in its general design the Type P engines, also on display, which have been on the market for some little time. Both models have aluminum cylinder blocks with steel liners, reminiscent of Caterpillar practice. Several of the engines shown were fitted with



Left: Armstrong-Siddeley "H800" Type HX 825 hp designed for Supermarine use. Right: also a Rolls-Royce H800, with new type aluminum cylinder linings

paired superchargers at the rear of the crankcase, with carburetors on the suction side. The new HX engine was so equipped. Incidentally it is rumored that the HX engine is to compete in the Schneider Cup Race, for which it will be "boosted" to develop some 1,600 hp. A. D. C. Ltd., displayed two very interesting new models—the "Hercules" 100 hp, air-cooled four-cylinder, and the "Aircore" 800 hp, six-cylinder in-line engine consisting of armoured cylinders and the "Norman" crankcase. This engine has a very interesting form of needle valve, shown in Fig. 1, by means of which it is claimed that perfect cooling is obtained. This company also displayed the very well known Napier water-cooled engine, a development from the old "Puma."

Two new models were displayed by the Armstrong-Siddeley Company, the "Major Gemini," a large edition of the Javelin Giant, with minor improvements such as inclined valve gear, and the enormous 14-cylinder "Laqueur" engine, announced some time ago, but shown in public for the first time at the Show. It is essentially a large edition of the "Jaguar," the details following very closely those of the smaller engines. The large diameter of this engine causes one to wonder if a "Vee" or "X" arrangement of cylinders would not be preferable for engines of such large displacement.

Other well known Armstrong-Siddeley models were present. Stay-in valve gear covers such the heat models. The method of supporting the roller arms has been improved on all models, and now gives a compensating action which minimizes tappet clearance wear.



Armstrong-Siddeley 14-cylinder 100 hp

constant with varying cylinder temperature. The fact that there were so many Armstrong-Siddeley at the Harlow display is all other makes put together leaves no room for confusion of these engines, and to the high esteem in which they are held by the Royal Air Force.

Attracting at all times a great deal of attention was the Napier racing engine developed from the "Loon" displayed in the Supermarine Race. This engine is advertised to develop 900 hp at 3,500 r.p.m., with a weight of 850 lb., and one was privately informed that it ran do considerably better than this when the engine demands. It carries a very interesting propeller reduction having double spur gears and axially arranged like the back gears on a helix. The gear is, of course, fully reduced and assists in the almost perfect streamlining to which this engine lends itself in the Supermarine racing plane.

The Standard Napier Loon engine was also shown in an improved model, with carburetor at the rear and

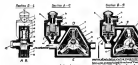


Fig. 1. Needle valve mechanism for A.D.C. "Aircore" engine. The 1. (left) shows the needle valve mechanism. The 2. (middle) shows the needle valve mechanism. The 3. (right) shows the needle valve mechanism.

located behind the cylinder blocks so that there can be axial streamlining. The number of speed and distance records held by the Napier Loon hardly needs mentioning. Major Segrave's car, which holds the world's fast road record, and also his motor boat which recently won the world's motor boat speed championship, were equipped with the racing model.

The Southern company showed the largest engine at

the Show—a new "544" (series III), 1,000 hp, 12-cylinder engine for airship use. The size of this engine can only be appreciated by actually seeing it. It is water-cooled with steel cylinder construction and semi-enclosed valve gear, and was developed from the type used on the successful trans-Atlantic flight of the airship R-34. The only Diesel engine shown was the Sunbeam Type P1, a 6-cylinder in-line engine of 100 hp at 1,500 r.p.m. This is a water-cooled engine apparently having steel cylinders with in-line aluminum water jacket construction. A totally enclosed fuel pump is mounted at the rear of the cylinder block. Very little could be learned about this design, and one wonders how far its development has progressed.

The De Havilland company showed its well known "Gipsy" 4-cylinder in-line engine, which is long manufactured in the United States by the Wright Aeronautical Corporation. The rating of this engine has recently been increased to 90-100 hp without increase in displacement. The Percival company showed one of their "Polaris" aircraft engines, a 6-cylinder radial of 60 hp, weighing 120 lb. including propeller reduction gear. This engine has been type tested by the British Air Ministry, and should form a very satisfactory power plant for light aircraft.

A. B. C. Motors, Ltd., displayed one of the enormous

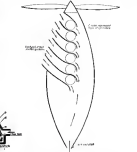


Fig. 2. (left) shows the needle valve mechanism for A.D.C. "Aircore" engine. The 2. (middle) shows the needle valve mechanism. The 3. (right) shows the needle valve mechanism.

ricious engines at the Show, the "Hercules," a 4-cylinder opposed type, to be mounted in the horizontal position it is rated at 75 hp, at 1,875 r.p.m. and weighs complete 225 lb. A 2-cylinder horizontal opposed model called the "Serpent," using cylinders and other parts interchangeable with the Hercules, was also shown.

Two small barrel type engines were exhibited, both of the single-ended type, air-cooled. The 90 hp "Redup



Aviad" has seven stationary cylinders of cast aluminum, while the Siatux, of Sivoi origin, is a rotary three-cylinder engine of 42 hp at 1,800 r.p.m. This engine also has seven cylinders, but in this one they are cast in one piece, and the sleeves form the cylinder barrels. Both engines are said to have been flight tested, but one is inclined to doubt whether either of them has actually reached a stage of development when it would be entirely satisfactory for general use in aircraft.

Both engines employ a "wobble-plate" in which the pistons are connected by connecting rods. In both cases a separate mechanism is provided to rotate the torque of the wobble plate. The Sivoi engine is made by the Reipur Lerer Engine Works of London and the other by Siatux Motor S. A., Zürich.

#### French Engines

A number of interesting French engines were exhibited, many of which are almost well known and require no detailed description. Among the French rep-



Above: A 6-cylinder Hispano-Suiza 600-hp engine. Right: The 100-hp Hispano-Suiza 100-hp engine. Below: The 100-hp Hispano-Suiza 100-hp engine. Below: The 100-hp Hispano-Suiza 100-hp engine.

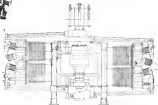
similar to those of Armstrong-Siddeley, was also on display.

The Société des Usines Renault showed a 4-cylinder vertical air-cooled engine of conventional appearance and of 80 hp. These and several other engines in the Show have evidently been inspired by the success of the British 4-cylinder inline engines. The Société des Moteurs Salomon displayed a few examples of its line of medium-powered air-cooled radial engines.

#### Italian Engines

Of the Italian exhibits the most prominent was that of Fiat, Ltd., which showed its 600-hp water-cooled 12-cylinder engine, and its 1,000-hp water-cooled engine of similar design. These have steel-water-jacketed cylinders, with inclined aluminum valve bearings. In addition to the water-cooled engine, there was exhibited a 7-cylinder air-cooled radial engine of 85 hp, which had aluminum cylinder heads with integral radial-arm boxes, characteristic of present American practice.

The Isotta Fraschini Company displayed the same line of engines exhibited at the New York show in February, including its 1,000-hp water-cooled tractor and the 12-cylinder air-cooled Van of 420 hp for military fighters. The 100-hp air-cooled 6-cylinder engine with aluminum cylinder construction should be mentioned



remains were Farman, who exhibited their 600-hp 16-cylinder high-speed water-cooled engine, which has already been described in the technical press, and a 250-hp 6-cylinder geared air-cooled radial engine having cylinders with integral radial-arm boxes quite reminiscent of American practice. An interesting feature of this engine, aside from the 2:1 reduction gear, was the fact that it was equipped with battery ignition. The Farman gear is now being built up because in a number of countries, outside of France, the American license being held by the Wright company.

The Société Hispano-Suiza exhibited its well known line of water-cooled engines, which presented their usual clean and compact appearance.

In addition to their long-line line of steel cylinder water-cooled engines, Lorraine-Dietrich showed a new 12-cylinder engine of very fine appearance, having three aluminum cylinder blocks and rated at 600/700 hp. This engine powered the French Air Ministry type test in June, and its appearance leads one to predict that it will be widely used in French military and commercial service. The company's line of air-cooled engines, quite

the cylinder axis. The valve ports were separate fixed aluminum castings accreted to the cylinder. The streamlined crankcase incorporated in this design has seemed an attractive possibility to the writer for a long time. It will be very interesting to follow the progress of an really streamlined radial engine. The "Colombo," a 4-cylinder vertical air-cooled engine of 85 hp, quite similar to the Clerus and Gipsy, was also shown in the follow exhibit.

#### Other Foreign Exhibits

Germany was represented by a number of engine manufacturers. The Aggas Motoren Gesellschaft exhibited a 90-hp water-cooled 6-cylinder engine of clean design. This engine mounts its two pistons in front on top of the two-ports, and carries its lubricating oil in a tank integral with the top cover of the crankcase. It is the first "wet sump" inverted engine the writer has seen. Mercedes-Benz, Ltd., exhibited its little 2-cylinder opposed engine, a bush has become well known on account of its use in the Klemin Daimler light airplane, and a model of its new 12-cylinder water-cooled engine of 800-1,000 hp fitted with overhead geared supercharger, built integral with the crankcase at the rear. This engine has roller bearings for crankshaft and connecting rods, one magneto and one battery ignition system, and an "Electron" (gasoline) crankcase. It is very interesting on account of the fact that it represents the re-entry into the large aircraft engine field of two of the oldest and best known aircraft engine manufacturers in the world, previous to the War.

J. Walter & Company, of Prague, exhibited its well known line of small radial engines in improved form, and in addition four new models. Three of the new models known as the "Vega," "Venus," and "Vixen," are of 5, 7 and 9 cylinders respectively, and rated at 85, 110 and 145 hp. They are all clean designs with accessories at the rear. The cylinders are of steel with aluminum heads screwed down when cold and held in place by split brass hold-on. It is claimed that the heads are easily removable without removing the cylinder barrel. The

valve gear, with its inclined rocker arms, is reminiscent of American practice, but the cylinder is definitely un-American by virtue of the very small angle between the valves. In this respect it resembles one of the Wright radial engines previous to the J-5.

The fourth new engine shown by Walter & Company, the "Castor" is a larger version of the trio previously mentioned. The cylinders are considerably larger than those of the other models, as the engine is rated at 240 hp with seven cylinders. The magneto are mounted "crosswise" just behind the propeller. This latter feature makes it less attractive in appearance than the other models. It appears quite conventional except for the cylinder construction, which is similar to that of the smaller engines.

#### Aircrafts

A number of interesting engine accessories were exhibited. The Société Renault showed two universal superchargers, one for exhaust-turbine drive and the other for gear drive with centrifugal clutch. The Junkers Motorenbau showed its interesting three-cylinder "Doka" fuel pump, a cross section of which is shown in Fig. 2. A very small and simple starter employing a blow-prover cartridge was shown on one of its German airplanes with a Gessert or Rhine "Twin" engine, and in the Italian exhibit was an air starter quite similar to the Bristol starter, which uses a very two-cycle engine to drive the air compressor.

The Victoria oil cooler mounting a Lanchester radiator appeared to be well worked out, and there were the usual number of exhibitors showing spark plugs, carburetors and so forth.

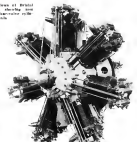
Two other accessories well worth mentioning were the Carter centrifugal oil separator, operated by the flow of oil to the tank and designed to be installed in an airplane as a permanent part of the lubrication system (The Carter oil separator is intended to improve the oil cooling—Ed.) and the Power Plus superchargers, which in an engine were type with the valves so arranged that it, do not lose against the walls of the housing.

among the conventional engines at the Show although it was exhibited in New York.

Another conventional Italian engine which attracted considerable interest was the 7-cylinder 90-hp Piaseolli air-cooled model designated as a "high altitude engine." Since it apparently had no supercharger, the high altitude feature appears to refer to the fact that the engine has a variable-pitch propeller, which seems to have a major part of the authority. One pointed out that the pitch of the propeller could be varied by hand at the will of the pilot, and as mechanism for preventing the abuse of this privilege with consequent over-loading of the engine could be discovered, at least without a better knowledge of Italian than that possessed by the writer. The spinner and crankcase of the engine formed a streamlined nose for an airplane, with all accessories mounted inside the streamlining at the rear. The cylinder construction was especially interesting, since the cylinders were made from split forgings with integral heads welded all over, with fore-and-aft valve seats set at nearly 45 deg from



Front and rear views of Bristol "Vixen" engine, showing oil and fuel air intake, carburetor and oil tank.



SPECIFICATIONS OF SELECTED *Airplanes* AND *Engines* DISPLAYED AT THE *Olympia Show*

[illegible]

*Note.*—Where two figures are given, the upper relates to the Upper Way (usually least turbid) form, and lower to Nephropsis. If only one figure is given, it refers to the Upper Way form.

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# THE 1929 National Air Races

## GET UNDER WAY

By HERBERT F. POWELL

**W**HILE WORKMEN were putting last summer's improvements on a somewhat damaged airport Cleveland's National Air Races and Aeronautical Exposition were inaugurated on August 24.

Infinite of both the brilliancy of the forthcoming 10-day aerial program and the vastness of the magnitude of twentieth century transport—on the show—a third parade, at more than an hour's length, passed through the downtown streets, terminating at the exposition building.

The face of this program typified the usual civic enterprises. Nevertheless it proved a wholehearted gesture on the part of the people of Cleveland in welcoming the gathering aviators. The passage through which 104 airplanes passed was highly and colorfully decorated, with an eye toward an aeronautical theme. A fleet of four Goodyear blimps, including the newly launched "Defender," were slowly scudded in escort. Significant prizes, one of which was refused at a downtown ticket office, suggested the spirit of the racing classic.

Proceeding formal opening of the six race program previous plans just completing a "Goodyear tour" of Ohio, moved over the city and landed at the airport. The exposition, held in Cleveland's 100,000 public auditorium, was officially opened at 1 p.m. by Leslie K. Bell, secretary of the Aeronautical Chamber of Commerce of America. In a dedication ceremony in the exposition building at 7:30 p.m. Mayor John D. Marshall and City Manager William R. Hopkins, presented the "key to the city" to Clifford Hendricks, managing director of the combined races and show.

Although heavy rain saturated Cleveland's airport on the day before the opening, weather forecasts indicated that stars would soak up the storm for the duration of the contests. In newly built roads leading to and from the airport, at the pile dress of the guards and official receiving buildings, in the uniformed and caped attendants, and in the condition of the airport itself, the visiting pilots, participants and spectators saw from Mr. Hendricks's well laid plans which began last February.

Sunday saw the beginning of the racing program and an attendance of more than 40,000 persons, which included governmental and civilian aeronautics officials. The first event on the card was Event No. 8, a civilian 50-horsepower open class with ONS engines. This was the first that some of the eight contestants tried the 10 mile pylons around the 5 mile pylons, the after was called off and postponed to later in the week. Douglas Davis and Arthur Chester, both flying ONS Travel Airs did some pretty flying on the pylons.



The winner of the Women's Air Derby, Miss Katherine Shadler, head and her fellow pilots across the Ford Run in the four women's derby race held in this country.

mph, and Link Phillips II. Love placed third with an average speed of 144.50.

Event No. 34, the next on the program, afforded the spectators the opportunity of viewing stunts "as the is done." It was the first of daily contests by these plane names "Foxy" Lord, "El" Henders, and R. W. Mackay, all of the Waco company, was Sunday's first money with a clever bit of stunt flying in Waco's powered tapered wing Waco. Every stunt magnitude was pulled by the two.

Although two Bachs were ready for Event No. 35, a 75-horse 15-day race for multi-engine planes, it was called off due to an insufficient number of entries, and Event No. 36 was moved up one on the program. This

event was the first of a series of "chase jumping" contests, the entries striving to land as close as possible to a circle marked out in the center of the field. First place honors went to L. C. Saylor, who landed 29 ft. from the center of the circle. Second place went to Edward Anderson with a distance of 156 ft. John P. Renner added an extra and unexpected thrill to the event when he missed the field entirely and landed on the avenue over the entrance to the grounds. In spite of his unusual descent, Renner was not injured. A total of thirteen jumpers "strayed off."

A dead-end landing contest was in order next, but there were only three entries. Therefore the entire "field" was "in the money," as it were. Orin Welch topped the biggest chase by bringing his Southern down to within 2 ft. 6 in. from the line. Ralph Winger was in bad luck, came to a full stop at 9 ft. 7 in. from the line, and before the shouting was over Larry Rich had halved his American Eagle 18 ft. 5 in. from the line. When race compares the distance differences of the three competitors it is perhaps just as well that there were no other entries, else the last place winner might have "freed landed" in the hole.

Following the "dead" event it was announced that the



One of the starting fields for the Cleveland air race pilots.

The nine "Black Bulls" took group of race pilots, shown right against in a formation. 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Orin Derby would take off. Six minutes took the air and eventually word was received that five of them had landed at Lima in the following order:

Vern Roberts, Monocraft with Vello engine; Lewis Lee, Davis monoplane with Latford engine; J. A. Spent, Barber monoplane with Genet engine; Dorsey Zimmerman, Barber monoplane with Latford engine; Leslie Miller, Barber monoplane with Latford engine.

The arrival of six Marine Falcons, seventeen Boeing Navy Fighters, eighteen Army Falcons, and the crews of the Air Corps (the Selkirk Field Corps) was the beginning of some famous street flying that brought back memories of the 1928 National Air Races.

As a means of proving that one can learn a radio plane in a hurry when one desires to leave, Elmer South of the entrance there, took off a Bellanca CH with six jumpers aboard. The six jumpers stepped off in a hurry according to schedule and the demonstration was a success.

The novel event of the day was the arrival of Hugh Johnson in a Waco with William Franklin in a soaring glider in tow. The pair had come all the way from Yakima, Mich., and when at an altitude of 3,900 ft. over the field, Franklin cut loose and glided down to a perfect landing directly in front of the stands. The landing was made in eight minutes from the "ten-four" to the final stop.

The feature event of the program on Monday was the arrival of the Women's Air Derby fleet completing the last leg of their race from Santa Monica. Louise McPherson

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in the CW class were: Phoebe O'Neil, Monocoupe; Mrs. Keith Miller, Kassar Fleet; Clara Poley, Travel Air; Tina Rusche, Del Moth; Bobbie Trout, Golden Eagle; Edith Feltz, Jayhawk. In the DW class were: Florence L. Barnes, Mary Van Mack; Hannah W. Noyes, Loane; M. Thaden, Oyd L. Kinn, and Harrel Croston.



B. H. Phillips (left) and M. V. Barnes, who are pushing publicly for the 1930 air races and trophies.

all Travel Air, Amelia Earhart, Lockheed; Ruth Nichols, Bearcat; Ken Roper, Navy Plane and Vern D. Walker, Curtiss Robin; Ruth Elder, Swallow; Gladys O'Donnell, Waco. Miss Harrel started a day late.

When the contestants arrived at San Bernardino for the first night's stop, Phoebe O'Neil led the CW class with an elapsed time of 32 min. 15 sec., and Bobbie Trout was second. Louise Thaden led the DW class with an elapsed time of 27 min. 30 sec., and Florence L. Barnes was second.

The stop for Aug. 19 was Phoenix, Ariz. Leaders were CW class, Phoebe O'Neil, 4 hr. 14 min. 15 sec.; DW class, Florence L. Barnes, 5 hr. 21 min. 19 sec. Edith Feltz and Gladys O'Donnell were second in the two classes. Clara Poley was out of the race at the point.

Between San Bernardino and Phoenix Miss Harrel Croston was killed when her plane spun into the ground. Three teachers saw the plane go down behind the hills, and it was some time before searchers reached the scene of the crash. The cause of the wreck is still unknown.

On Aug. 20 the fliers reached Douglas, Ariz., with Louise Thaden, 5 hr. 33 min., and Phoebe O'Neil, 6 hr. 24 min. 27 sec., leading. At El Paso, Tex., on Aug. 21, the same two led with elapsed times of 6 hr. 48 min. 31 sec., and 8 hr. 35 min. 26 sec. Amelia Earhart was second in the DW class, and Edith Feltz second in the CW class.

At Fort Worth, Tex., Aug. 22, Louise Thaden, 11 hr. 4 min. 30 sec., and Phoebe O'Neil, 13 hr. 28 min. 30 sec., remained clear. The cause of the wreck is still unknown. On Aug. 23 the fliers reached Dallas, Tex., with Louise Thaden, 5 hr. 33 min., and Phoebe O'Neil, 6 hr. 24 min. 27 sec., leading. At El Paso, Tex., on Aug. 21, the same two led with elapsed times of 6 hr. 48 min. 31 sec., and 8 hr. 35 min. 26 sec. Amelia Earhart was second in the DW class, and Edith Feltz second in the CW class.

The last overnight stop before the arrival at Cleveland

was made at Port Columbus, the municipal airport at Columbus, Ohio. Mrs. Thaden and Mrs. O'Neil were still ahead, with elapsed times of 19 hr. 15 min. 40 sec. and 24 hr. 14 min. 32 sec. Miss O'Donnell was still second in her class, but Tina Rusche had moved up to second place in the CW class, while Bobbie Trout was listed to drop out and the times of Mae Harrel and Edith Feltz were not figured because of irregularities in their schedule.

The final standing of the contestants in the DW class as they arrived at Cleveland were: Louise McPherridge Thaden, first; Gladys O'Donnell, second; Amelia Earhart, Hannah Noyes, Ruth Elder, Nina Paris, Mary Van Mack, Oyd L. Kinn, Mae Harrel, Vern D. Walker. In the CW class the order was: Phoebe O'Neil, Tina Rusche, Mrs. K. Miller, Edith Feltz. All five of the leaders in the DW class had Whittaker-Sund planes, while Mrs. O'Neil's plane had a Warner engine.

A bit before the arrival of the Women's Derby Verna D. Croston took first honors in the daily closed stick landing contest. Another would not do in the case of Croston, as he landed right on the line.

The arrival of derbies being in order, the next event was the arrival of the All Ohio Derby. M. A. Speed in his Great powered Barling monoplane took first place money with a total elapsed time of 4 hr. 22 min. 50 sec. Lewis Lewis finished second with a total elapsed time of 4 hr. 43 min. 15 sec., and Barney Zimmerman took third place money with a total elapsed time of 5 hr. 13 min. 1 sec.

The first closed course event was an affair that has been an important part of the Air Races since the early days. . . . the Liberty Eagle Builders Trophy race. Maj. J. Sidney Owen flying a Curtiss Falcon was first place winner with an average speed of 143.07 m.p.h. for the 120 miles 12 lap course. Second place went to Benjamin Sowd who flew his Curtiss Falcon over the course at an average speed of 133.89 m.p.h. Third place was captured by John Gill who maintained an average speed of 132.72 m.p.h. Seven pilots entered the military observation plane race and six of them were outclassed. The feature of the race was the dense trees which, on the advice of Major Owen. On one lap he let it up to 150 m.p.h.

When the smoke had cleared away from the Liberty engine race the daily jumpers went down to step off and make down again. Joseph Crane landed 22 ft. from the center of the circle and walked away with first place honors. Edward Anderson landed 60 ft. 4 in. from the "spot" to win second place and L. C. Saylor came down 200 ft. from the spot. When one compares Monday's distances with Sunday's one gets the impression that the boys are getting farther and farther away from home.

Soon after the chairs had been folded up and the boys off the field, Lee Schoeninger set his snappy looking Waco powered Lockheed Vega down on the field to take first honors in the one-stop derby from Los Angeles. Schoeninger's time for the 2,072-mile plane was 15 hr. 51 min. 10.8 sec., or an average speed of 149.65 mph had he been on to New York at that speed he would have kept up a new trans-continental record.

The attendance at Monday's meet was estimated at around 50,000 people. The mayor from Calumet and Mrs. Lindbergh will be present brought on in special motorcade. The military planes did their usual setting up exercises and to add color to the programme, four Goodyear Shags maneuvered about the sky.

# INSIDE THE Exposition Building AT CLEVELAND

By LESLIE E. NEVILLE

ALTHOUGH the airplane and engine exhibits were fewer in number than those of some of the previous exhibitions, the representatives at the National Aeronautical Exposition on the opening day (Aug. 24) presented an excellent cross section of the industry. Of

the 20,000 persons who witnessed the day's activities at the airport, it is reported by officials that at least one-half of that number attended the exhibit at the Cleveland Public Auditorium during the first day and evening.

In striking contrast to some of the previous exhibitions, ample space was provided for visitors to stand off and view airplanes from a distance. The idea of leaving or roping off exhibits seems to be growing in popu-



The interior of the Exposition building as it appeared on the opening day.

larity as several of the manufacturers employed this means of presenting an attractive exhibit. However, this method has the disadvantage of preventing many interested persons from getting a close view of the product being shown.

ENVELOPE EFFORT was made by the show management to provide comfort and additional entertainment for visitors, and loud speakers were installed in all parts of the hall for the purpose of broadcasting music and announcements. In addition to this, several musical features were presented on the steps of the arena and broadcast to the more remote portions of the exposition building. Demonstrations were simple but effective and consisted mostly of airplanes and other forms of motors with landing in the national colors. In one portion of the exhibit a very appropriate form of decoration in the form of a number of colored silk wind cones, blown by electric fan drafts, was employed.

Of the 54 airplanes in place on the opening day, only a few were new designs. Most of the others were developments of models previously exhibited or adaptations of existing designs to new power plants. In the latter case the tendency toward the provision of greater power to provide adequate reserve was almost always the case. Two of the newest designs had not yet been test flown but were ready to complete for the show and shipped by rail. These were the General Airplane Company "Mushpiles" and the Great Lakes Aircraft Corporation's twin-engined amphibian.

The Mushpiles, which embody a number of original features, is an open cockpit sesquiplane powered with a Series C Pratt & Whitney Geared Hornet engine and having an all aluminum monocoque design. This plane has a normal gross weight of 6,250 lb. and a pay load



The Curtiss "Thrush" as interesting part of the exhibit at Curtiss Flight Center.

of 2,000 lb. The Great Lakes amphibian is a biplane type powered with two American Curtiss engines having a total horsepower largely of metal.

Among the other new designs were the Ercato Curtiss two-engine tandem side-by-side sesquiplane, a light three-engine four-seat and training purposes, and the McCord "Quadrant," a two-engine cantilever monoplane amphibian powered with two Wright New Whetstone "Five" engines mounted in nacelles slanted into the upper surfaces of the wing. The wing is of wood and plywood construction, the skin being of duralumin half-plywood. The hull is of composite wood and steel

construction and the amphibious landing gear is hydroplaning only retractable.

Several interesting developments were seen in the Boeing Division of the exhibit of United Aircraft and Transport Co. In the main hall were shown the Boeing Model 100, which is a commercial version of the Wasp powered military machine produced in last company and the 40-104 a five-passenger cabin biplane for either mail or tourist transport use and having a convertible passenger-mail compartment. The 16-passenger transport Model 80-A monocoque plane was shown on the grounds, between the auditorium and the arena. The commercial version of the Yonker "Coronet" was also shown in United Aircraft and Transport.

A Great Lakes training plane and a Tiger Wing Whiplane were suspended in various attitudes of flight from the ceiling of the main hall.

The new Curtiss Thrush, a folding wing cabin sesquiplane bearing some resemblance to the Robin, and also powered with the Challenger engine, was shown in the exhibit of Curtiss Flying Service.

A three place, biplane sesquiplane of conventional construction and powered by a Wright J-5 engine, was exhibited by the Marshall Aircraft Corp. and a four-engine monocoque powered with a Le Rhod 60 engine and having side by side seating arrangement was displayed for the first time by Samuel Tashman. The Le Rhod powered Island Sport sesquiplane and the Golden Eagle Chel also were displayed for the first time at the show.

AS IS USUALLY THE CASE, the engine exhibits were well arranged and consisted to most cases of displays of completed engines and parts to show the type of construction. A New Whetstone Nine, with a perfect cut away to show the operation, having an electric motor keeping it in constant motion, was shown by the Wright Aeronautical Corporation, which also displayed its other radial airplanes and the Wright Gyro. Geared and direct drive models of the Series C Wasp and Hornet engines were shown by the Pratt & Whitney Aircraft Company.

Much attention was attracted by the Curtiss "Crusader," a six-cylinder-in-line, inverted, air-cooled engine rated at 120 hp. This engine probably marks the first attempt to combine air cooling and inverted cylinder arrangement in a six-cylinder engine. Cooling is accomplished by means of an air scoop located on the exhaust side of the engine and leading air from the upper part of the cylinders. The air shaft, which is called an "under heater" type by its designers, because of its position below the cylinder head, is driven from a vertical shaft at the rear of the engine and of the engine.

A six-cylinder radial engine having three valves per cylinder and other unusual features was exhibited by General Aeronauts Company, Inc.

Many types of accessories were shown and a number of these will be described in a later issue.

Airplane manufacturers and others will be interested to know that the Hill Van Breda Metal Works, of Greenport, Ohio, is offering ready made cooling of the N.A.C.A. type as well as other forms of cooling and metal shape.

Also exhibited in the necessary exhibits were a number of Micron units by the Westinghouse Company and an airplane engine supercharger by General Electric Company. The Goodwin "Air Wheel" was also exhibited. A valuable addition to hangar equipment in the form of a power unit designed to cool and eliminate dusts of larger doors, was shown by Allen & Drew, Inc.

## THE S. A. E. PROPELLER AND POWER PLANT SESSIONS

By EDWARD P. WARNER

ENLIGHTENED for the aeronautic standard session which was an initial feature of every S. A. E. aeronautic meeting, the engineering discussion in Cleveland opened with a propeller session on Monday afternoon, followed by a meeting on power-plant topics in the evening. The general tone of the propeller session, where three papers were presented, was addressed rather to the adjustment between the propeller and the engine, and to modifications in the propeller which would make it possible to maintain good engine operating conditions at all times, thus to theories of propeller design.

The afternoon session had to compete with the Air Races as an attraction and the attendance was reduced to about 50 as a result. A large proportion of the propeller experts of America was present, however, Lieut. Carl B. Hager of the Navy, who was to have presided at the meeting, was called away and Maj. Leslie MacLean, lately commanding officer of Wright Field, took his place.

Two papers were contributed on variable pitch propellers, one by T. F. Wright, chief engineer of the Curtiss company, and W. R. Turrell, inventor of the Turrell propeller and now consultant for Curtiss. The other author was Frank W. Caldwell, formerly in charge of the propeller branch at Wright Field and now chief engineer for the Standard Steel Propeller Company. Both papers entered into extended discussion of the general advantages of a variable-pitch propeller without regard to any particular mechanical construction. Mr. Wright and Mr. Turrell called attention in particular to British computations showing that for a typical piston plane and an underspeeded engine the ability to vary the propeller pitch at will would increase the rate of climb by 30 per cent at sea level and by approximately 75 per cent at 10,000 ft. The maximum speed in level flight would be unchanged at sea level, but increased by some 30 per cent at 20,000 ft altitude. The authors especially stressed the benefit to be drawn from the varying of pitch at high altitudes, even without supercharging. It is in point that is frequently overlooked.

It has been commonly assumed in the past that the ideal performance with a variable pitch propeller would be secured by holding the engine rpm to a constant figure for a given throttle setting at all times. Recent tests made by the Curtiss company and first presented in the Wright-Turrell paper, emphatically disprove that, so far as economy is concerned. Economy studies were

made with a Curtiss "Robin" recently carrying a propeller of 5 ft. pitch and cruising most economically at about 76 m.p.h. and 1,480 r.p.m. An increase of the propeller pitch to 8 ft. loaded the engine down so that the same cruising speed was secured at 1,350 r.p.m., with a resultant reduction of approximately 30 per cent in fuel consumption when adjusted for the best mixture for economy. Curve given in Fig. 1, reproduced from the paper, shows the relationship between economy and propeller pitch at various speeds and carburetor settings.

The results of the test indicate that in cruising it is advisable to use an even higher pitch than in level flight at maximum speed. The advisability of atmosphere that without manual control was gained out as a weakness in the aforementioned proposed type of variable pitch propeller such as the Hebe-Shaw, in which the governor operates to maintain constant r.p.m. for any given throttle setting.

THE ATTENTION also enlarged upon the possibilities of a variable pitch propeller for improving the take-off and in making it possible to get the full benefit from a supercharged engine. Estimates of performance for an airplane having a ceiling of 30,000 ft. without supercharging showed climb to 20,000 ft. in 32 min. with a supercharged engine and a fixed-pitch propeller, reduced to 20 min. or a gain of 17 per cent by the ability to vary the pitch. The variable-pitch propeller was also shown to have special advantage on the multi-engine machine, an adjustment can be made to maintain the full efficiency of the surviving propellers in case of the failure of one engine. The propeller on a dead engine, too, can be set for a 90 deg. sweep back angle so reducing the parasite resistance by an amount which the authors estimate as 40 ft. for a 9 ft. propeller at 60 m.p.h. With such variable pitch propellers as far used in flight it would be impracticable to make the sagittal drive large enough to permit of folding the blades in the proper position, so the control has been through linkages, built in the design upon which Mr. Turrell has been engaged the drive is directly through gears and there is no strain to the extent of extreme of the blades.

Mr. Caldwell and the authors of the other paper agreed in holding themselves to be the best material for a variable pitch propeller on present-day engines. Mr. Wright suggested, however, that steel has promising possibilities, especially in very large designs.

Both papers devoted some attention to a recapitulation of the history of variable-pitch design and of its present status. The operating mechanism of the propeller designed by T. A. Debs for the Standard Steel Propeller Company is shown in Fig. 3, taken from Mr. Caldwell's paper. It provides for manual control through a linkage, and for a throttle stop to prevent passing the blades through the neutral position with the throttle open and damaging the engine by over-speeding. The Turnbull propeller, so which the other paper expressly refers, has been in use for many years of development behind it in approximately its present form. It embodies a drive by electric motor mounted on the shaft and geared down by some 60,000 to 1, so that when the motor is running the blade angle is being continuously varied at the rate of one degree of angle every two or three seconds. No throttle stop is now provided, but it is pointed out that electrical control permits of the ready incorporation of any such self-governing devices. Parenthetically, it appears to the present writer that neither the manual control, nor the constant slow changing of angle as long as the clutch is engaged or the engine switched on, nor the automatic control by governor accords the ideal. There is an opening for further inventive effort in developing a follow-up gear which will make it possible for the pilot to set a blade to any desired blade angle, the blades then being shifted exactly to that angle by mechanical or electrical power within 5 or 10 sec.

On the design described by Mr. Caldwell no definite figures were given, but Messrs. Wright and Turnbull indicate that the Turnbull type has been brought down to a total propeller weight of 100 lb for a 200 hp engine. Neither paper made much of the possibility of pitch reversal as a desirable point, although Mr. Caldwell's comment on it briefly. When Caldwell's work with variable-pitch propellers first began, some 11 years ago, the possibility of reversal to secure braking action in landing was highly concerned in one of the chief aspects of pitch reversibility, but the introduction of wheel brakes has changed the outlook.

Mr. Caldwell's treatment of the subject took up more detail on laboratory results than did the comparison paper. A group of test results taken with a propeller approximately 9 ft in diameter indicated that a reduction of blade angle at 4500 rpm from 22 to 14 deg would increase the static thrust by 40 per cent. The prospective improvement in take-off is obvious.

A point in variable-pitch propeller design often overlooked is the variation of stress conditions with changing

angle. Mr. Caldwell treated this in some detail with special reference to the bending and twisting moments produced by the centrifugal force on the blade. A curve of normal moment from this source, reproduced in Fig. 2, shows not only a great variation in magnitude but an actual change in sign of the moment within the probable range of angular adjustment of the propeller blades. The general method of stress analysis described by Mr. Caldwell is an simplification of the classically simple one given in most available texts on propeller design. In particular Mr. Caldwell recommended careful allowance for the distinction between the minor principal axis, the neutral axis, the chord line of the blade section and the plane perpendicular to the line of applied bending moment in the section, instead of treating them all as identical as in the simple theory. His stress theory also allows for the reactive bending moment produced by centrifugal force acting on the deflected blade, and calls for a calculation of blade deflection, not necessarily made in connection with wind propellers.

The availability of the conditions drawn on the two papers is interesting, especially since Mr. Caldwell and Mr. Wright are respectively the chief engineers for two companies that have had a large production of fixed-blade metal propellers, but neither of which has as yet actually put a variable-pitch design on the market.

Mr. Wright and Mr. Turnbull, of course, naturally the next step will use the statement use of the controllable-pitch propeller on airplanes of many types and on practically all airplanes of certain types." Mr. Caldwell concludes: "It is dangerous to say that the aerodynamic advantages of controllable-pitch propellers are becoming increasingly important to propeller-pitch speeds and engine horsepower increase. With improved materials and design, we should soon be in a position to meet the practical requirements of the problem."

In the course of an active discussion of the two papers, Commander Havel, former head of the propeller branch of the Naval Bureau of Aeronautics, and now with Bendix Aviation Corporation, took up the cause of the propeller of essentially variable pitch, especially for small planes. He suggested that the feathering of the propeller on a stopped engine, the use of the variable-pitch propeller with a reverse pitch as a brake source of major importance and that the purpose of pitch variation in flight required a range of angular adjustment of only 5 to 10 deg. Commander Havel described a device for feathering variable-pitch propellers designed for the Navy which is expected to give a variation of 3 deg of blade angle with an increase of weight of only 8 lb over the fixed blade propeller. The design is that of Victor Lockhead and bearings of the blade are substantially strengthened by the negative device of floating the blades and carrying the centrifugal forces upon them on a large number of glass wires in tension anchoring the blade roots to the hub.

Mr. Caldwell expressed the belief that there was no definite limitation upon the use of manual control for varying the pitch and that it could be worked even with engines of very high power, but that a pure speed-sensitive control would prove simpler. The writer contributed to the discussion, remarks along the line of those that have been interpolated in this report. Dr. Max Munk, lately in charge of the aerodynamic test work of the National Advisory Committee for Aeronautics, and now with the Alexander Lippincott Company, discussed in connection with the present type of wind tunnel test of airfoil models in a basis for propeller design. He indicated

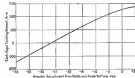


Fig. 1—Change of induced moment at root of blade of adjustable-pitch propeller with change of blade angle setting.

briefly the results of recent researches of his own showing the true lift coefficient for a section as a part of a propeller to be 40 per cent higher than for the same section tested as an airfoil, the difference being attributed to the uneven conditions of airflow about the propeller blade. The acceptance of this view would fundamentally modify the basis now commonly accepted for propeller design.

The first feature of the propeller section was a paper in which T. P. Wright, who had a hand, this time in collaboration with R. E. Johnson of the Curtiss company's aerodynamic laboratory. The subject was "The Gearing of Aircraft Propellers."

Following upon a recapitulation in general terms of the gain in propeller efficiency to be had by increasing P/N/D through gearing, which reduces the rpm more than it increases the diameter, the authors gave specific examples of certain existing prop and installations. One particular case was that of an Army observation airplane designed around the Liberty engine. When the Liberty was replaced with a more modern type, giving 2 per cent more power with 10 to 15 lb less weight, but rated at 2,300 rpm instead of 1,800, the maximum speed dropped approximately 4 mph. The same explanation was found in a drop of propeller efficiency of about 4 per cent, corresponding to a 20 per cent reduction in P/N/D.

Another point which involved less attention, but upon which the paper commented at some length although with an unfortunate absence of numerical illustration, is the effect of a small or large blade behind the propeller on its efficiency. Gearing down the propeller, with an increase in diameter, gets the blades away from the interference region and reduces the loss from this cause. Mr. Wright and Mr. Johnson suggested that the effect must be an important one, because gearing has in some cases produced benefits distinctly too great to be accounted for by the variation of propeller efficiency alone.

Another argument for gearing was found in the increased rotational speed and power of aircraft engines, which has now reached the approximate limit of safe absorption by a direct-driven propeller, even though it be made of metal. A tip speed of 1,000 ft per sec was taken as representing the limit of good practice with metal blades, and it was shown that it would be impossible to keep them anywhere near this figure with a 600-hp engine turning a propeller at 2,800 rpm. The maximum permissible power point at this rate of rotation and at 130 mph can in fact be calculated from data given in the paper to be only 265 hp. Of course in some cases tip speed is allowed to exceed the figure just specified.

The disadvantages of weight and cost in a geared installation were frankly recognized. Fig. 4, taken from

the paper, shows the relationship between the weight added by gearing and the horsepower for a number of engines. While the chart shows a wide variation among individual designs, the average weight increase will be seen to be about 0.2 lb per hp, which seems to fit in with a very moderate price in pay for the advantages secured by gearing in many cases, especially on large and slow-flying planes. Although the lowest weight per horsepower for any of the gear installations included in the chart is that of the Napier Lion racing engine, Americans will note with satisfaction that the four best figures are those of American engines and all are stock types, not special racing designs.

On the heading of cost the authors referred to the geared and ungeared Conquest engines as showing the extra labor and material necessary for geared drive. The same illustration also shows the advantage of gearing in entering the stall on the approach and indicating a better stream-line of the nose of the fuselage or nacelle, especially when the engine is not to be run reversed.

The loss of power due to friction in a gear drive is also given to be overestimated. Mr. Wright and Mr. Johnson indicate that it should be kept down to about 2 per cent in a year gear drive, and 4 per cent with an epicyclic. Balancing advantages and drawbacks against each other, they came finally to the conclusion "In general, gearing should be done when the weight and cost of engines rated at less than 400 hp, or for airplanes weighing less than 4,000 lb. Airplanes having design values of P/N/D = 0.7 or greater, and at the same time having a propeller tip speed less than the limit given, 1,000 ft per sec, for material and 880 ft per sec, for wooden propellers, would not give an increase of performance of sufficient magnitude to warrant the added complications. However, when the designed performance of an airplane does not come within the range of values stated, a designer should be hesitant about installing a direct drive."

Reducing-gearing is a refinement of design and, very often, a similar gain of performance can be obtained by using more care in attaining clearance of the ground-clearance characteristics of a given airplane. In the past, given the same engine, the correcting gear has been instead of giving super-performance to an originally well-designed design. This fact should be realized by the aircraft designer before he finally decides on gear pinning. So far as consideration on any engineering problem in performance, we believe that the truth in large main engine transport airplanes will be readily available to the use of gearing."

In the course of the discussion of this paper, Major McColl referred to the relation between gearing and cost and to past experiments with gear reduction which had not been successful in the airplanes because of the increased weight and area of the medium required with a larger propeller and lower disc-area velocities. He warmly expressed the importance of gearing, however, and expressed personal conviction that a had not unduly stressed in American literature the gains of the Allison Company challenge. Mr. Wright's conclusion that engines of low power should not be geared. He had found it advisable to gear them in many instances. It was his opinion that the weight of gearing depended upon all the engine characteristics and especially upon the torque characteristics and not merely upon power. Mr. Wright further expressed surprise at the mechanical efficiency given for gear drives. He

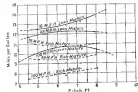


Fig. 3—Change showing relation between propeller pitch and that induced at various angles. Standard propeller pitch unit is 3 ft.

himself, believed that 90 per cent should be possible. Mr. Insley of the Continental Engine Company expressed a doubt of the validity of any great relief about the cause in which getting should be used, a doubt with which other speakers in the discussion agreed. He tended to concur in general with Mr. Wright's view about the susceptibility of air to small engine.

The power plant section, which was presided over by Capt. L. M. Workman of the Fordham company and attended by about 100 members and guests, included an extended treatment of the work done on high-temperature liquid cooling by the Military Division of the Army, Air Corps. It was presented by Gerhardt W. Frank, of the Power Plant Branch at Wright Field, where he has been in active charge of the recent studies in that field.

Mr. Frank pointed out that the general suggestion of using liquids of higher boiling point than water for cooling cylinders has frequently been made, in the past. The first serious work was obtained when the work was done up at McCook Field in the spring of 1923 at the suggestion of S. D. Hanna. Mr. Hanna also proposed the trial of ethylene glycol, which has been used ever since. Laboratory experiments quickly showed such promising results that the first flight trial was made a little less than a year after work in the laboratory began. Studies have now covered three types of engines: the Liberty, the Curtiss D-12 and the Curtiss V-1570.

Owing to the great, or too, superabundance of a chemical description and a good cause for the liquid used in the cooling and to the incomplete reproduction and inaccurate interpretation that Army releases on the subject have occasionally received, the exact nature of the cooling fluid used has at times been noted in misinterpretation, and the impression has been made to have been that the liquid used was a new invention. Mr. Frank made it clear that this was not the case. He pointed out that it was discovered some 70 years ago, that it is one of the oldest gases and that it has a boiling point of 367 deg. F. and freezes at 11 deg. F. Commonly enough, however, its addition to a little water greatly lowers the freezing point and the mixture known as Freonite used instead of pure ethylene glycol in the Army Air Corps work was used by Mr. Frank to maintain about 2 per cent of water and have a boiling point of 350 deg. F. and a freezing point of 0 deg. F. When the ethylene glycol mixture has been used in all of the world, so far, other liquids are also under investigation at Wright Field.

Laboratory tests with a standard engine and Freonite used for cooling have shown, according to data drawn from curves given in Mr. Frank's paper and reproduced

in Fig. 5 herewith, very little effect either on brake horsepower or on fuel consumption as the temperature of the cooling liquid at the outlet from the cylinders is varied from 160 to 300 deg. F. The power dropped about 2 per cent, while the fuel consumption only having increased about 4 per cent at an intermediate temperature, fell back to its original level. A serious point revealed in the paper, however, Mr. Frank did not comment upon it in detail, was the difference between the temperature of cylinder heads and barrels with water cooling and when cooled by Freonite at the same temperature. In the cylinder heads, this difference was nearly 80 deg. in favor of the water cooling, the same presumably being a difference either in the heat conductivity of the liquid or in the turbulence of flow through the system and the amount of entrapped air that dry pump or in conditions of differing surface transfer of heat between the liquid and the metal.

The change of temperature in the liquid itself during the extent of the cooling system of course depends primarily upon its rate of flow. With the same pump used in both cases, the range of temperature in Freonite was about 35 per cent higher than water because of its lower specific heat.

The next step, as described by Mr. Frank, was to go on to flight testing, and a power airplane was modified to reduce its radiator outlet and terminal area to the highest cooling temperatures presented. It was found



Fig. 4—Increase of engine weight by addition in cylinder

possible to reduce the amount of cooling surface of the radiator by 70 per cent and the volume of water carried by 40 per cent, with a total reduction of about 100 lb. in the weight of the cooling installation. Mr. Frank described as troubles as having occurred in any of the two, except for wear and sticking of piston rings and the loss of some of the cooling liquid through the various joints. There have been no difficulties so serious as to interfere with continuance of the tests.

The most recent development described by the author is the use of ethylene glycol for cooling a high-compression engine, using a special fuel and taking full advantage of its anti-knock properties. The compression ratio of the D-12 engine used in the test was raised to 7.5, and the fuel employed was California gasoline with 11 cc. of ethyl lead per gallon. The effect of the changes has been to raise the power of the engine from 425 to 2,400 r.p.m. to 520 to 2,600 r.p.m. and to reduce the fuel consumption from about 0.45 lb. per horsepower-hour at the most economical operating setting to 0.40. Mr. Frank summarizes the combined effects of high-temperature cooling and the use of a high compression ratio made possible by special fuel in reducing the power-weight weight per horsepower from 2.00 lb. to 1.46 lb. The

data given in his paper make it impossible to show by further comparisons that the aggregate frontal area of engine and radiator per unit of power would be reduced by about 25 per cent, while the complete installed weight including fuel and oil for 4 hours running at 85 per cent of maximum speed, would be cut approximately from 3.83 to 2.61 lb. per hp. The only disadvantage in efficiency to set against these advantages would be the loss of propeller efficiency due to increase of rotational speed which was analyzed in the paper of Messrs. Wright and Johnson.

The Frank paper was followed by a prepared discussion by Mr. Goswami who described work done by the Navy along the same line. The speaker's attitude toward high temperature cooling was considerably more optimistic and on the basis of his experience in the Navy he was even inclined to think the radiator could be eliminated entirely the cylinders being exposed to the air, and all surplus heat disposed of directly to the air from the surface of a steam-blast water jacket possibly arranged to insure maximum surface. Mr. Goswami declared that Ethylene Glycol could give more uniform cooling than air and predicted that liquid cooled engines would now be brought back to a strongly competitive position with the standard type.

In response to a question, the author of the paper stated that no trials had been commenced with the installation of solidified parts in Freonite cooling systems. Mr. Goswami of the Army Company suggested that good fuel economies obtained with Freonite at standard compression ratios were probably due to reduced mechanical friction losses at high temperatures. The discussion was concluded by Samuel D. Hanna, of Wright Field, with the observation that experience had shown that engines with high temperature liquid cooling would stand, without complete collapse, those that could be quite disastrous in the standard type. During the whole course of the discussion, it was evident that the question of the development of a new type of engine, those engaged in aircraft engine work, and there were several remarks upon the availability of ethylene glycol cooling for a Diesel engine.

Mr. Frank's paper had been preceded by one on "Gearing Engine Performance," by J. H. Goswami, superintendent of design of engineering of the Great Engine Corp. Mr. Goswami's object was to provide analytical comparisons between engines of varying performance qualities to determine how particular conditions in design would affect and overall usefulness. After comparing assumptions that such analysis had been so common in the past, the author first indicated the analysis of the effect of engine weight on operation and, concluding the engine weight to be changed without change of any of its other qualities, and the power to be increased enough to keep the pay load of the plane constant.

Assuming a engine life of 3,000 hr. and a total life of 6,000 hr. for the airplane the effect of increasing the engine weight from 2 lb. per hp. to 2.47 lb. was calculated to be an increase of \$250 in total operating expense during the life of the airplane for a 300 lb. engine. Mr. Goswami's calculations relate to his engine having the same weight as the previous one, but with the additional weight of the heavier engine, in this instance, per unit increased cylinder volume making it possible to run at a higher compression ratio with lowered volumetric efficiency and more effective pressure.

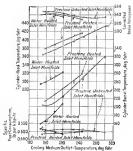


Fig. 5—Effect on engine performance of change of outlet temperature of cooling

was. The high engine, weighing 2 lb. per hp. was figured for a compression ratio of 5 and a brake mean effective pressure of 120 lb. per sq. in., while the heavier standard design was to have a compression ratio of 6 and a brake mean effective pressure of 98 lb. In this case the heavier engine came out the better of the two on total operating cost, showing a saving of \$350 during its useful life. This was due almost entirely to lowered fuel consumption due to the higher compression ratio.

The author also pointed out that even having the fuel consumption entirely out of consideration, the increase of engine weight could be used to secure increased durability which would make up for the cost of carrying the extra weight around. His final conclusion seemed to be, although not explicitly so stated, that the saving of weight in engine design has been overdone as an objective and that lower fuel consumption, greater durability and better overall economy might be obtained with somewhat heavier construction. He closed by repeating a suggestion previously made that the power plant weight per pound of net thrust was much more important than the engine weight per horsepower and he expressed the hope that the users of engines would give more careful attention to ratios of paying their true economic worth an aim which he did not believe to be accomplished by the present criteria of engine performance used generally accepted.

The discussion of the paper was brief, but served to develop a marked difference of opinion as to who should be responsible for engine installation and cooling design. Mr. Goswami and several members of the audience believed that cooling should be furnished as part of the power plant by the engine manufacturer, while the airplane designers preferred to keep it in their own hands. In reply to questions, Mr. Goswami said that he would use high mean effective pressures in his designs only if forced to do so by what he believed to be the most and if informed public demands.

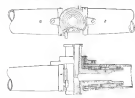


Fig. 6—Liquid-cooled engine with radiator and cooling system

## FIRST FLIGHTS OF THE

## Metal-Clad Airship

By JOHN T. NEYILL

**W**ITNESSED by several thousand persons, including many of the best known naval and civil aeronautical engineers at the nation, the Detroit Aircraft Corporation recently sent down their metal-clad dirigible, ZMC-2, "a distinctly American airship," which they built for the United States Navy.

The craft's initial flight at this writing has been succeeded by a second one, and the builders hope to have attained by the time this is published more than half of the 30 hours of flight required before delivery to the Navy.

Both flight tests thus far made, although primarily in nature, seem to have satisfied the engineers' expectations as to the point to which they have been carried. That to the

fact that the craft's 202,000 cu ft capacity metal hull has been only partially inflated for the flights, and because the helium in the bag is yet to be run through the ducts, no definite figures on the ship's lift can be procured at this time. Official performance figures, it was said, will be issued by the Navy Department after the ship's arrival at Lakehurst.

How closely the ship's lifting performance will come to the company's forecast figures, however, can be judged from the following estimates based upon the two flights already made: The ZMC-2 was designed to have a gross lift of 12,242 lb., (at 100 per cent inflation with 92 per cent pure helium at 60 deg. F.) 3,442 lb. of which is useful load. In the initial flight it has been estimated that the hull carried approximately 580,000 cu ft. of gas and that it lifted between 2,300 and 2,350 lb. of useful load, this load being made up of five men, 100 gal. of gasoline, 400 lb. of water, 150 lb. of oil, five parachutes, and 140 lb. of mail. On this basis it has been estimated that, when fully inflated with gas testing

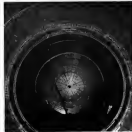
approximately 99 per cent pure (such as that now in the hull), the useful load will be increased to approximately 2,950 lb. With further elimination of a major portion of the carbon dioxide and water now in the helium, the useful load is expected to rise to the neighborhood of the figure originally forecast.

The process of filling the metal hull with helium was carried out in the following manner: The two balloons were first filled with CO<sub>2</sub>, after which CO<sub>2</sub> was forced slowly into the bottom of the hull. Due to the tendency of the gas to freeze, the connections were heated by torches. Throughout this process a head of one in. of water was maintained in the hull by restricting the exhaust openings, excessive pressure being taken care of by exhaust valves in the balloons. The air was exhausted through the top of the ship to a point outside the hangar until the exhaust line showed 99 per cent pure CO<sub>2</sub>.

The helium line was then erected and helium admitted at the top of the ship through three manholes. Exhaust gas was run through four inflation sleeves in the bottom of the hull and carried outside the building. Purge streams were taken at various points in the ship and in the piping to determine the stratification and composition of the mixture. When the exhaust gas showed about 40 per cent before the exhaust was connected to the purifier. The air was first forced out of the purifying plant by the exhaust from the hull. The exhaust was passed through a pump in the bottom of the scrubber, caustic soda solution being circulated through the scrubber and the tanks. In passing through the scrubber the gas mixture came in contact with the caustic soda, which absorbed most of the CO<sub>2</sub>. After 99 per cent, or better, purity was found in the scrubber exhaust the exhaust was introduced in the hull in a point approximately 95 ft. above the bottom. To compensate for the volume of CO<sub>2</sub> absorbed in the scrubber, helium was added from cylinders atop the hull. When the desired purity was obtained the CO<sub>2</sub> in the balloons was replaced by pure helium, filling the ship ready for test flights.

**N**O ARRANGES yet been made to try out the ship's speed. On flights already made it was said that the craft's top 220 hp. Wright engines were turning over at speeds ranging up to 1,200 r.p.m. and that the air speed registered up to 47 mph.

Despite its size (the fineness ratio being 2.88) the ZMC-2 seems to have little of the picking qualities that might be expected in an airship of its size and shape. The riding tendency is even less noticeable



Interior of the big dirigible under inspection. An extension ladder is shown reaching up to the upper portion of the structure.

Carl D. Pritsche, vice president of the Detroit Aircraft Corporation, who was a passenger on the ship's baptismal flight, stated that the metal hull seemed to dampen rather than amplify the engine exhaust notes. According to Capt. W. E. Keyser, the ship's test pilot, the new method of control gives the ship a high degree of maneuverability and excellent maneuverability. vibration, he said, also was noticeably absent.

Beside Mr. Pritsche and Captain Keyser, the initial flight carried Master Sergeant Joseph Hudak, Keyser's assistant, and Edward J. Hill and Arthur G. Scholten, D.A.C. engineers in charge of the dirigible's construction. Lieut. Comdr. William K. Harrell, and members of the naval board of survey, designated to inspect the ship, and Comdr. Garland Fulton, chief of naval aircraft construction, witnessed the flight from the ground. Comdr. Fulton participated in a flight the preceding day.

Before formal delivery to the Navy is made, the dirigible will undergo a number of required tests, including a 15 ft. per sec. ascent under power, a 15 ft. per sec. descent under power, descent as a free balloon (200 ft. in 4 min.). Time, gas and air tightness, speed and reliability will be vital factors.

From its engineering standpoint, the ZMC-2 introduces three major new lines of thought, the first being its all-metal construction, the second its entirely new system of control, and the third its unusually low fineness ratio. The ZMC-2 hull is 149 ft. 5 in. long and 32 ft. 8 in. in diameter, giving it a fineness ratio of 2.83. It has two balloons, the forward one having a displacement of 22,000 cu. ft. and the rear one a displacement of 28,000 cu. ft., the two comprising approximately 25 per cent of the hull volume. The balloons are made of two-ply reinforced fabric weighing 9 oz. per sq. ft. The seams are lapped 2 in., reinforced and taped but not sewn. Unlike other airships the ZMC-2 carries its lifting gas directly within the hull itself, rather than inside one or more balloons located within the hull. The balloons in the metal-clad dirigible are inflated with air to regulate the gas pressure.



The metal-clad ship after ZMC-2 is suspended in the state of its first test flight over Detroit.



One is immediately impressed with the simplicity with which the ZMC-2 is constructed both inside and out. By the sheer strength and workability of metal the builders have done away with the multi-layer and complex joining systems prevalent in modern fabric covered dirigibles. The ZMC-2 has but one skin, and that, at



Typical girders used to stiffen the interior of the metal hull

"Aluminum," so that the company was compelled to invent a new automatic riveting machine, to "sew" the pieces together. The skin's gauge is .0085 in., and it is not corrugated.

For gas-tightness the builders developed their own laminating sealing compound, which was applied under the seams immediately after the riveting had been completed and approved.

The ZMC-2 was built in two vertical halves, the framework of the halves progressing from the nose and stern towards the middle. Briefly, the dirigible's framework consists of 24 longitudinal, of single rolled channel sections and five main transverse circular frames of built-up longitudinal section, both being stamped out for lightness. Although seven frames of similar section are spaced between the main transverse members

to avoid the thin skin as reinforcing the craft's shape, usually all of the tensile stress is carried by the five main rings and longitudinals. Two of the innermost main transverse members are used to support the ship's bow and control surfaces, which are placed 30 ft. forward of the tail. The control car, located midships, is attached to the central main transverse rings. The fifth main ring is located well forward to take care of stresses on the nose. Forward landing gear are attached to this member. All of the framework is of duralumin, the thickness of the longitudinals varying from .014 in. to .012 in., according to the intensity of compressive loads in any particular part of the hull.

Instead of the usual control surfaces near the tip of the tail the ZMC-2 has eight fins behind the hull 30 ft. forward of the stern. The ship has four rudders, one attached to each of the four vertical fins. Similarly, two elevators are operated off the four horizontal fins. Total fin area is 4.00 sq. ft., total elevator area, 190 sq. ft., total rudder area, 95 sq. ft., and total automatic rudder area 55 sq. ft. All of the fins and control surfaces are similar in construction to those of the Ford all-metal monoplane, and all were subjected to wind-tunnel tests. In placing the control surfaces well forward of the tail the designers hoped to achieve the smooth flow of air in motion as feasible in the zone where the hull has its greatest diameter. It is stated that the tests have shown that this construction favors non-stressness in the controls and gives the ship a high degree of maneuverability. Controlled air's extent from the air-lanes so perfect through metal tubing led to the hull and to the control car when one wheel takes care of the rudders and another operates the elevators. Like the control surfaces, the control car is covered with corrugated duralumin, the metal being of much larger gauge than that of the hull. Besides a large window extending across its front, the car has a window and a door on each side, as well as a window at the back. Gasoline is carried in a 200 gal. water tank, for ballast, is also located in the rear compartment.

Figures of the ZMC-2 are carried on rubber strappers, one on each side of the car, and in line with the

AVIATION  
August 31, 1939

AVIATION  
August 31, 1939

side windows. Each power plant has an independent fuel, oil system and control system.

One of the most interesting features of the Detroit Aircraft Corporation's dirigible lies in its unique method of construction. It has been previously stated that the hull was built in two upright halves, after which the halves were placed horizontally and riveted together. Metal plating girders, the hull was cut in strips from 4 in. to 17 1/2 in. wide, and, necessarily, had to be bolted with just the proper length and curvature to line up with no adjacent strips in order to allow not the slightest separation from the hull's vertical and platform contour. Beginning with end rings at the nose and at the stern, these strips were riveted in the last piece by piece and the nose or stern, as the case may be, bolted toward the larger ring as the work progressed. In order to join these thin pieces with a minimum showing a minimum of deformation, an automatic riveting machine was used.

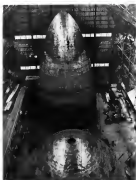
From a mechanical viewpoint the construction of the dirigible, as Mr. 1939, parallels the major project and is expected to become a valuable contribution in the mechanical



Front view of the car, with engine units and machine in place

and construction in metal aircraft. The machine is capable of spacing, inserting and completing 155 rivets per minute, at about 40,000 rivets in one 8-hour shift. The cost per thousand feet of the skin is declared to be about the same as that of sewing, connecting and taping one foot of fabric seam. The automatic riveting machine is set on a carriage, mounted on concentric rails placed directly under each half of the hull. The machine frame is adjusted to the proper angle of the hull plating, a gauging device on the machine giving a correct and uniform overlap of the plates. In its operation three strands of wire, one .035, one .012 and .008 in. diameter and three rows of rivets are spaced simultaneously. The machine closes off the wire, rivet length, the wire wires are punched through the two sheets of metal after which a revolving arm heats up the rivets. With this machine, two men are able to accomplish as much work in a given time as 1.28 men working by hand.

Over a 60 day period more than \$300,000 was spent



The completed halves of the ZMC-2 in the position in which they were riveted, before being slipped over and joined

in engineering research and study in developing the ZMC-2, previously, all of this before Congress approved and the \$300,000 (paying the way for the Navy) contract. To date, according to the builders' own statement, the ship has cost about \$250,000. The experimental work embraced the detailed designing of a number of hypothetical metal strips of various sizes. And as each square foot of hull surface a thorough stress analysis, including the compression of all factors, was worked out. A metal water model, built to 1:12 scale, was constructed and tested at different angles of pitch to verify the theory of the design, as well as the effects of the main bending and shearing stresses and the safety features involved.

It is anticipated that all-metal construction in dirigibles of larger size does not increase the weight as compared with fabric construction. This is due to the fact that the metal skin carries the shear stresses and a certain portion of the tensile stresses, as well as reinforcing the internal fabric members, thus reducing the quantity and area of internal framing necessary. The company's research work, in which they had the co-operation of the Bureau of Standards, the Massachusetts Institute of Technology, the Aluminum Company of America and several additional agencies, furnished the following conclusions:

That the new hull form and fin arrangement constitute an improvement in aerodynamic qualities. That the results of the large-scale metal water-model test, checked with D.A.C.'s stress formulae, gave practical proof of a remarkable gain in strength obtained with the metal-plate design. That difference of the metal hull, tested and doped in the company's furnace, was only 1/10 lb. per

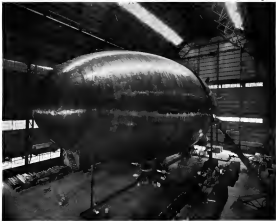


Automatic riveting machine used to "sew" the skin of the ZMC-2. Note how from which rivets are formed bending over members of the structure. Riveting work is shown at left

sq. meter in 24 hours, or much less than with the best future comparisons. That the thin sheeting and seams have a satisfactory resistance to vibration.

In fairness to the builders, it should be remembered that they make no claims of commercial value so far as the ZMC-2 is concerned. Rather, they designed and built it as an experiment to prove the new aerodynamic and structural theories a refinement, and to prove the adaptability of metal to its deformed purpose. The claim is advanced, however, that the metal-clad ship, constructed on a larger scale, will have a definite commercial and military value. In connection with the paper work done by the ZMC-2's engineers in 1922, it is noteworthy that the completed craft's empty weight, 18,000 lb., exceeds the detailed estimates made at that time by only 100 lb., despite the substitution of Alclad for duralumin about fifteen per cent thinner.

By way of history, it might be noted that the ZMC-2 was conceived by a group of Detroiters in 1911, when, in 1922, organized the Aircraft Development Corporation. Harold H. Ericsson, who had had charge of airplane engine building for the government during the World War, was president of the corporation, and Carl B. Fritzsche, vice-president and general manager. Ralph H. Upson was the company's first chief engineer, and he has gone a large share of the credit for the design. Costly and heart-breaking experimental work



This ZMC-2 needs two workers with helms cylinders cut out and bent to shape



Directing the first members of the crankcase of the ZMC-2

consumed the first four or five years. A Navy contract was signed in the fall of 1926, and actual construction of the present experimental ship was begun immediately thereafter. The change to Alclad was made a little less than two years ago. Just recently the Aircraft Development Corporation became the nucleus of the Detroit Aircraft Corporation, operated on a wider field. Edward S. Egan is president of Detroit Aircraft Corporation and Mr. Ericsson is chairman of the board.

## OVERHAULING

# OX-5 Crankcase AND Cylinder ASSEMBLIES

BY JAMES P. WINES

**T**HE RIGID inspection and maintenance system as found at Parks Air College, Inc., East St. Louis, Ill., for the purpose of keeping the OX-5 parts in the school training planes in operating condition and the methods employed in performing major overhauls on the crankcase and cylinder assemblies were discussed in the first two articles. In this, the last of the series, the processes in the overhaul of the accessories and the system employed in setting in the engines after overhaul will be taken up.

The carburetors of the OX engines are practically rebuilt by the Parks service organization "Murphy" Shellenbaker, who is the service manager, realizes that the improper functioning of a carburetor may nullify the results for a forced landing; and while the rebuilding entails considerable expense, Murphy reasons that the added cost is more than wiped out by the inherent danger of forced landings. In fact, he says that the school never has engines which are in such good condition in all respects that "they won't quit" regardless of the cost of overhaul and upkeep.

The first step is the overhaul of a carburetor, of course, is its complete disassembly. The seat of the needle valve is then examined, and is replaced if it is found to be worn in the slightest. New weights, new pins and a new needle are always installed. Incidentally, it is standard shop practice at the Parks school to secure the pins in the

*The Last of a Series of Three Articles About the Service Organization of Parks Air College*



A section of the motor service department at the Parks school

weight assembly by soldering them. The first chamber cover is replaced if it is worn where the needle passes through, while the float shell is always replaced, if it is old, without regard for its condition. The replacement of the float naturally saves the Parks organization a great deal of trouble. Next, the collar on the needle valve is located by measurement, and the float assembly is put together.

The carburetor body is then inspected for wear, and the butterfly valves and the shafts are checked. These parts are replaced if they are worn perceptibly. To insure a richer mixture when idling, so that the engine will not cut out as in sometimes the case when the throttle

is fully closed a  $\frac{1}{8}$  in. hole is drilled just above the delivery valve in each barrel. The examination and replacement if necessary of the high speed compression and spark plugs comes the next step. This is followed by the installation of speed belts with one inch drop below the jaw, instead of the usual two. These belts have been found to give a great deal of "grit," as the danger of water in the gasoline is minimized by their use. The exhaust tubes are next plugged with cork, and are fastened in place with wire ties.

Mr. Stencheski says what is quite true, that there is no point in keeping the strictly adjustment of a plane equipped with an OX-5 engine seldom gives sufficient trouble to make use of it. The majority of the air born and the check, starter follows as the next step. The distributor is first placed on one of the overhead engines and is checked through-out the operation of running the engine in. It might be added that in the overhead of a carburetor if any of the jets on the floatless rise are found to be broken, they are unscrewed with a screwdriver and are jarred according to the U. S. 808 formula. The new portion is then rechecked and the holes are refilled. Also, the setting of the valve on the needle valve is checked by hand test before final assembly so that the gasoline level is  $\frac{1}{8}$  in. below the top of the high speed jet.

The water pump on an engine goes through overhaul at the Parks shop. Likewise is completely torn down for inspection. New bearings for the drive shaft are always used, while the other parts are replaced only if they are

worn. The bearings are bored to allow a clearance of .002 in. after the diameter of the shaft has been accurately measured with the use of micrometers. Each ply of .002 in. on the shaft, which is regulated by the motion of special machines, is also allowed. After this work has been performed the pump is regulated and assembled. All the holes but one are plugged up, and the pump is tested for leaks by attaching a water hose.

IN OVERHAULING one of the Berling magnets with which most OX-5 engines are fitted the first step is that of testing it on the Weidenshoff stand which is a part of the shop equipment. In this test, the condition of the condenser, the winding and the magnets is noted. The shaft is also checked for end play. The magnets are then run down, and the commutator and brushes are tested separately. The high tension parts are next tested by means of a high tension coil.

Following these tests, any faults that have been discovered are corrected at once. The bearings are then inspected for wear and cracks, and if suitable for further use are retained. If not, new bearings are substituted and the magnets are assembled. The bearings on the assembly operation are packed with a mixture of vasoline and Mobilie "16" engine oil. It has been found that this mixture need not be harder as do some lubricants, although it is necessary to loosen it up with two or three drops of oil after every 15 hr. of operation.

The next step in the overhaul of a magnet is per-



An aerial view of the Buick Airport showing the hangars, aircraft and other buildings.



Below—The possible damage that could result from test run the Weidenshoff test stand. Above—The Weidenshoff test stand used to determine condition of magnets, condenser and condition of magnets.



haps the most interesting. The distributor gear is set in relation to its normal position. Then, a new Bosch "217" breaker assembly is taken, and the taper is changed to fit the shaft of the Berling armature after which it is mounted with the breaker mounting screw. The arm on the regular Berling breaker box is then moved off, while the arm hole is matched to slip over the advance stop pin of the Bosch breaker assembly. By this means the box is locked in the proper position. Next, the breaker points are adjusted, leaving a clearance of .018 in., and the position of the distributor pencil set. No. 1 electrode is checked with the breaker points just opening. This is to make sure that the pencil is over at the time the points close.

At the conclusion of the work the magnets are again placed on the Weidenshoff test stand, which has a gap assembly, and a break-down test is made. It might be noted that the Weidenshoff test stand, like the Strom connecting rod boring tool and auger, is a highly priced piece of equipment in the Parks shop. It will not only test almost every type of magnets, but it has a con-

struction for testing battery systems quickly. The device is used, not only in performing general overhauls on the magnets, but for testing them as a part of the regular 100 hr. inspection of the engines and planes. Mr. Stencheski says the reason Parks Air College has but one forced landing on an average in every month period because of magnet trouble, is the result of the use of the Bosch breaker and this test stand.

At any rate, a magnet that has been overhauled is placed on the stand and is run at from 500 to 3,000 r.p.m. for some minutes. During this run, the bearings are checked to see that they are not overheating while a check is also made to see that there are no leaks in the electrical circuits. If the magnet throws a maximum hot spark for 15 min. and no end play has developed, it is returned to the engine from which it was removed. The engine is then fired.

With the replacement of the magnets, the wiring is hooked up and the engine is ready to go to the test stand. The stand is used at the Parks school is particularly interesting, since it is an uncovered framework of milled steel, constructed by the students in the welding shop. It has a landing gear and tail skid. A fuel tank and a radiator are located just back of the engine mount, and behind this is an instrument board, having a tachometer, a water temperature gauge and other necessary instruments. An open switch with a porcelain base is used as the ignition switch, while the throttle control is carried back to approximately its normal position in reference to what would be the pilot's cockpit. On both sides of the fuselage, supports have been arranged to hold heavy planks which extend from the nose to a point behind the instrument board. These planks act as rails for the next in change of running the tests on the overhauled engine.

THE ADVANTAGE of the fuselage test stand, though, lies in its portability. When an engine has been overhauled and is ready for test, the stand is wheeled into the shop and there the engine is mounted on the bearings. The gasoline and water connections are made, as well as those to the instrument board and switch. A standard Hatzel wooden propeller, raised at a climb, is mounted on the shaft, and the fuselage is wheeled slowly to start the test. When the stand is in the proper position, a

bucket filled with sand is attached to the tail to hold it down, chains are placed in front of the wheels, and a ground rail, constructed of wooden strips and wire function is placed in front of the propeller. The rail runs to the ground. When the training in process has been completed, the fuselage may be wheeled to any place where the newly overhauled engine is wanted, leaving a great deal of unimpaired landing. Another thing to be checked, of course, is the oil, when the engine is tested. In this, Mr. Shostetshin points out, it is highly advantageous to have the engine mounted in the position in which it will be installed in a plane.

An overhauled engine is first run at from 300 to 500 r.p.m. for about three hours, or until the water temperature drops to 140 deg. F. The throttle opening is then advanced by any stages until the engine is turning over 800 r.p.m. It is operated at this speed for three hours if the water temperature does not exceed 150 deg. F. Next, the engine is given sufficient "heat" so that it turns up 1,000 r.p.m. In many cases, the water temperature will jump almost immediately to 180 deg. F. If this happens, the engine is throttled down until it runs at 140 deg. F. and the throttle opening is slowly advanced until the engine will run at that temperature at 1,000 r.p.m. It is kept at this speed for three hours. Following this, it is operated for one hour at 1,100 r.p.m. another hour at 1,200 and one-half hour at 1,260 r.p.m. The engine is then idled for five minutes, after which the throttle is opened for ten minutes as it would be normally in flight. The next five minutes, the last in the test, are run at full throttle. If the engine turns up 1,400 r.p.m. with the Hartzell propeller, while the water temperature remains at from 150 to 180 deg. F, it is approved after it has been fired over more and tested.

**A**NOTHER IMPORTANT PART in the fabrication of an engine, during the process of running it in, and also in its operation later in a plane, is the use by the Parks aviation school of what is known as "jerk" service. This has not yet been placed on the market commercially, in fact, it was just by chance that the Parks school came across it, so to speak. Mr. Shostetshin says the service, a Mr. Shaw, who lives across the river in St. Louis, came out to the Parks Aviation day and announced that he had an oil line relieving valve system so that they would not stick. The oil, he insisted, would not burn, and, consequently, would have no carbon deposit. It was to be mixed with the gasoline in proportions of one ounce of the oil to every five gallons of fuel.

Mr. Shostetshin agrees that he was skeptical of the inventor's statement. However, they conducted a test by pouring a small quantity of Upper Cylinder Oil and a small quantity of ordinary engine oil into a pan. An attempt was then made to start both engines. The engine of himself, that the Upper Cylinder Oil would not. As a further test, two mixtures of the cylinder oil were placed in the fuel tank of one of the OX-5 powered training planes together with five gallons of gasoline, and the plane was taken up for a hop by one of the pilots. Mr. Shostetshin said that the reason for using double the amount of cylinder oil required was to determine its value and value. When the plane was landed, he made an inspection of the valve stems and guides. The stems, he says, had the stems that all aerobically hope to keep, but that none ever does. As a result, one ounce of Upper Cylinder Oil is now used with every five gallons of gasoline that go into one of the Parks Air College en-

gines. In addition, a quart of Upper Cylinder Oil is mixed with every three and one-half gallons of crankcase oil when the engine is run in.

**T**HE COST OF OVERHAULING an OX-5 engine at the Parks school varies from \$180 to \$350, depending, of course, on the condition of the engine when it is received at the shop. While the costs vary to this extent, Mr. Shostetshin believes that a profit could be made if outside work was accepted and a charge for overhauling an OX engine of \$225 was made. In the several articles, mention was made of the fact that the Parks aviation organization, with its staff of eight mechanics which is augmented by the employment of advanced students in the mechanics course, is overhauling the large number of OX engines now on hand at the school as rapidly as possible. The purpose of this is to provide spare engines for the school training planes, to provide engines for installation in the Parks P-1 biplanes, which are soon to be placed in production by Parks Aircraft, Inc., and to provide a sufficient number of engines for the transportation of an overhaul service for the owners of planes powered with OX engines.

According to the present plan, the engine service will be similar in some respects to that which it is possible to obtain on automobile batteries at the present time. That is to say, the owner, whose engine needs overhauling, will fly his plane to the Parks Airport. The plane will be wheeled onto a barge and in just the time that it takes to jerk one engine and install another, he will be on his way again with a newly overhauled OX-5 engine. The engine which he flies away is his. For the \$225, or whatever the standard charge may be, and his old engine, the ownership of the overhauled power plant will have been transferred to his name. The old engine will then be overhauled, and, in turn, will be transferred to someone else.

It is argued that this type of service will not appeal to a plane owner who has a "good" engine in his plane. However, Parks officials feel that once the standard of the service work becomes known, there will be no reluctance on the part of the owners in parting with their engines. They argue that an engine overhauled by their service organization is even better than a new engine. Incidentally, the reason the plans for the engine service are being laid out is to require that the engine, which is ended in, be flown to the field by the owner in his plane, in so make sure that the engine is in operating condition. That this is essential may be seen by the comparatively low price for which Mr. Shostetshin believes that the service can be rendered.

It is not known just how soon the Parks organization will be able to inaugurate its engine service. In accordance with the Department of Commerce regulations, Parks Air College will have to maintain an overhaul shop solely to render the work required on the engines and planes used in training. That will mean the formation of a separate unit to handle the service work, and an enlargement of the mechanical staff. It is known that the service will be inaugurated as soon as possible, that certainly the Parks organization hopes to have service stations at the principal airports throughout the country, and that Mr. Shostetshin, as service manager, is now attempting to develop tools for overhauling radial air cooled engines with the same degree of accuracy and efficiency now evident in the work performed on the OX-5 power plants.



## "Sun God" Flight A Distance Mark

**Minor Covers 7,200 Mi.  
In Avoiding Repeating Trip**

**SPOKANE, (A. M.)**—Breaking all distance records for sustained flight, the "Sun God" plane, piloted by N. W. Smith and Earl W. Dyer, was launched last after completing a one-stop trip across the continent and back in 115 hr. 53 min. 30 sec. The field airplane, powered with a 300-hp. Wright F-5 engine, was released as dawn broke above the city of Oakland, Calif., while Bernard Cheyenne, Wyo., Smith's pilot, was based at Pullen Field, Wash. (Cleveland, N. York, Cleveland, St. Paul, Aberdeen, S. D., Boise and Missoula, Mont.). A final demonstration of endurance was made before the plane was launched at Pullen Field and the pilot signified their willingness to continue the trip by making the National Air Corps Association, which sponsored the attempt, landed that single proof of the practicability of refueling on cross-country flights had been given and that there was no need to continue.

Approximately 7,200 mi. were covered at an average speed of 67.2 mph, despite loss of time in refueling, reaching the airport at Rock Springs of one night in seven, and a similar delay at the airport of a storm near Boiseville, Pa. Only 22 hr. 38 min. were required on the return trip from the time it departed New York to the time the plane arrived over Spokane, making the average speed close to 100 mph. Considering the conditions of the journey, it is noteworthy with the record of 19 hr. 10 min. 42 sec. for a flight from New York to Los Angeles, according to Ray Cregor, manager. The Associated Chamber of Commerce has sponsored the flights in a class of expense.

The 7,200 mi. covered between most and points, Spokane, Idaho, made no use of the airplane record of 4,700 mi. made by Farnham and Del Pratt, but the 4,200 mi. covered by the OX-5. Despite an on its return trip from Pullen Field to Idaho, it came to a landing without any incident.

The trip was covered not by plane from Missoula Air Transport, and also handled by Texas Oil Company. Texas' aviation plane and engine of were used throughout the trip.

## American Indians Tour Stops

**INDIANAPOLIS, (A. M.)**—Plans for the first American Indian tour have been announced. The first will start here Sept. 16, cover a route with overnight stops at Chicago, Terre Haute, South Bend, Fort Wayne, returning Sept. 28. It is expected that between 20 and 30 planes will start.

# GENERAL NEWS

## Clarence M. Young Nominated As MacCracken's Successor

**Major Will Arrive  
New Duties on Oct. 1**



Old Clarence M. Young

**WASHINGTON, (A. M.)**—Clarence M. Young, Director of the Civil Aeronautics Branch of the Department of Commerce, has been nominated by President Hoover to succeed William F. MacCracken, Jr., as Assistant Secretary of Commerce for Aeronautics. Mr. MacCracken's resignation was before the President on Aug. 15, and will take effect Oct. 1.

After being notified of his nomination, Major Young made the following statement: "I thoroughly appreciate the honor which the President is conferring because the position of Assistant Secretary of Commerce for Aeronautics is an exceedingly important one. Civil aviation in the United States has made tremendous progress during the past three years under the wise direction of Secretary MacCracken and it is hoped and planned that this same constructive leadership between the aviation industry and the department of Commerce will continue for many years."

## Manufacturers to Select Own Detroit Show Space

**DETROIT, (A. M.)**—Manufacturers are to have the opportunity to draw lot and select their own space in the exhibition building for the Detroit Motor Show, a privilege for the first time. This is a new development after the 31st Detroit Motor Show, which was held at the Detroit Convention Center. The Automobile Chamber of Commerce has sponsored the show.

Manufacturers of the Automobile Chamber who have had space applications were scheduled to choose their respective locations. The floor plan presented at a meeting of the commercial manufacturers' section of the Chamber of Commerce during the National Air Races. Mr. Cregor announced that 12,500 sq. ft. of space has already been sold.

Detroit next year will still be close to a new \$1,800,000 municipal exposition building now being erected in the City Airport. The building will be 1,000 ft. long and 200 ft. wide, affording a clear area, except for one line of stands, for the display of the cars. The building on the airport will be from 100 to 150 ft. wide, allowing entrance of the largest transport planes without dismantling. The building will be of brick and metal, with red stone trimmings, and may be finished by December.

## Developed Aeronautics Section

He was appointed Chief of the Air Engineering Division of the Aeronautics Branch, Sept. 1, 1935, in which capacity he built up the aeronautical inspection force which now carries on licensing, identification and inspection of civil aircraft, and also licensing of pilots and mechanics and the enforcement of air traffic regulations. He also directed the final preparation of the Air Commerce Regulations which were promulgated as the Federal Air Regulations, No. 1, 1935.

He was appointed Director of the Aeronautics Branch, July 3, 1937, and has served in that capacity for two years. He is a member of the Yale Club of New York and Washington, and of the Society of Aeronautics Engineers.

Mr. MacCracken, the retiring Assistant Secretary of Commerce, is (Continued on Page 483)















## Foreign News Briefs



The appearance of a Boeing 707 at one of the Mexican airports.

(Continued from page 475)

is powered by Rolls-Royce engines with a new type compressor. The engine cylinders are of alloy. New air compressors, developed since then at the 550 hp in the Rolls-Royce "JL" from which it is designed.

The jet took to the air, the first of all, emerging from the engine through a cone in the shape of ribs on the sides of the fuselage to the fin and back to the engine through ducts in the bottom of the fuselage. Unlike its predecessor, the "JL," the machine has not wings, but back into the wings of engine and the radiator. The fuselage, built in metal and the metal skin is designed to take most of the stresses.

Rollers are provided here to the effect that at least two of the rollers are provided with two engines of 1,000 hp each in tandem arrangement. The propellers of the rollers have been streamlined with air-foil shape. The rollers are of the type, it is known that the first rollers and the rollers have been built in the shape of a cone. The rollers have been built in the shape of a cone. The rollers have been built in the shape of a cone. The rollers have been built in the shape of a cone.

## Propose India Mail Flights

RANDOLPH (Indon)—Arrangement for a landing field here and a Mail-India (Indon) Airlines, Inc. and the Indian Airlines are being made to accommodate the movement of the Indian air mail to Singapore. The field is expected to be built near the Indian Airlines landing field. The Indian Airlines are being made to accommodate the movement of the Indian air mail to Singapore. The field is expected to be built near the Indian Airlines landing field.

## French Air Carriers Viable

PARIS (AP)—A French airline is expected to be established in 1960. The airline is expected to be established in 1960. The airline is expected to be established in 1960. The airline is expected to be established in 1960. The airline is expected to be established in 1960.

## American Opens Air Line in Mexico

MEXICO CITY (AP)—American Airlines announced today that it had secured a mail and passenger service between Mexico City and Los Angeles. The service is expected to be established in 1960. The service is expected to be established in 1960. The service is expected to be established in 1960.

The scheduled flight was for 1,000 air mail and 500 passengers. The flight was for 1,000 air mail and 500 passengers. The flight was for 1,000 air mail and 500 passengers. The flight was for 1,000 air mail and 500 passengers. The flight was for 1,000 air mail and 500 passengers.

## French Line in S. A. Extended

BUENOS AIRES (AP)—The French Line's new air service to South America is expected to be extended to include the city of Buenos Aires. The French Line's new air service to South America is expected to be extended to include the city of Buenos Aires. The French Line's new air service to South America is expected to be extended to include the city of Buenos Aires.

A new airplane factory is being built near the Villahermosa Port of Mexico City. The factory is being built near the Villahermosa Port of Mexico City. The factory is being built near the Villahermosa Port of Mexico City. The factory is being built near the Villahermosa Port of Mexico City.

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Another step in preparation for the proposed Franco-Brazilian Airline is the construction of a new airport near the city of Rio de Janeiro. The new airport is expected to be completed in 1960. The new airport is expected to be completed in 1960.

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## WHAT OUR Readers Say

## Landing Speeds Again

To the Editor:

In the issue dated July 20th, 1959, Aviation, you published a letter from me, Mr. E. J. G. and the point brought out is that the landing speed is not the same as the take-off speed. The landing speed is not the same as the take-off speed. The landing speed is not the same as the take-off speed.

In large commercial airplanes, the landing speed is not the same as the take-off speed. The landing speed is not the same as the take-off speed. The landing speed is not the same as the take-off speed.

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ship is usually decreasing and the load is increasing. The landing speed is not the same as the take-off speed. The landing speed is not the same as the take-off speed.

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the engine times the weight of the engine in the air. The landing speed is not the same as the take-off speed. The landing speed is not the same as the take-off speed.

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## THE BUYER'S LOG BOOK



## American Transport Goggles

FOLLOWING a long period of development the American Optical Company, South Bridge, Mass., is offering the American Transport Goggle to meet the needs of airline pilots. This type of goggle was designed primarily for the United States Army. The ventilating system and the lens design as well as other features, are the result of more than two years of experimentation.

Two separate rubber cushions, which can be removed and replaced, and which fit closely to the contour of the face but do not interfere with nose breathing, are provided. The distance between lenses is freely and permanently adjusted without tools with a rubber locking device. All parts are replaceable and interchangeable because the goggle is built to exacting specifications.

On the down curve of each eye cup is located a tube which catches the slipstream. As the air passes through this tube it creates a suction behind the lens and ventilating air is drawn up through intake valves at the bottom of the eye cup. The constant ventilation behind the lens is provided to prevent fogging. Openings are staggered so that no direct air "blows" or "rushes" reach the face regardless of the speed of the plane or the velocity of the slipstream.

Lens acquisitions were made by Dr. E. D. Volper, chief of the scientific staff of the American Optical Company. The lenses are said to be free from aberration or distortion and designed so that the optical center



The American Transport goggle showing rubber lining behind the nose piece.

is moved to a position in direct line with the straight ahead line of the vision. They will not cause goggles headache and provides for complete eye comfort. The lens can be supplied in white glass or in colored glass and is patented. Colored glass is a glass reducing variety which does not alter color values.

## Aviation Trophies

AVIATION trophies and prizes of all types are now being offered by the Aero Specialty Company, 1168 So. Hill Street, Los Angeles, Calif. The company is at a position to furnish these trophies according to the requirements of the purchaser. Also included in this line are a number of novelties such as airplane's radiator emblems. These emblems are carefully reproduced models of well known airplanes.

## Consolidated Tachometer Shaft

ANNOUNCEMENT has been made by the Consolidated Instrument Company of America, Inc., 305 E. 43rd St., of a new type of tachometer drive shaft known as "All Bronze". This new shafting, a patented product of the company, consists of an inner cable connected of 17 strands of high carbon, high tensile



Photograph of the "All Bronze" tachometer drive shaft made by the Consolidated Instrument Company.

strong music wire in four layers under special process. The casing, a bronze interlock aluminum type, is reinforced with copper braided aramid, is very light in weight. It has a natural bronze finish with no painting or lacquer to scale off. It is also very flexible and will permit of sharp bends making installation handling easier.

The company also is now manufacturing a full line of aircraft and engine instruments and is also in a position to furnish weather instruments for installation at airports and meteorological stations.

## Fafnir S Series Bearings

AS A result of an extended study of the special bearing requirements of the aerospace industry, the Fafnir Bearing Company, New Britain, Conn., has announced a new series particularly adapted to these requirements.

These bearings, known as the S Series, are of the single row radial type with such dimensions and narrow widths. They are available in sizes from  $\frac{1}{16}$  in. to 1 in. in diameter.

Bearing No.	Bore (Inches)	Outside Diameter (Inches)	Width (Inches)
51	$\frac{1}{16}$	$\frac{1}{8}$	$\frac{1}{8}$
52	$\frac{1}{8}$	$\frac{1}{4}$	$\frac{1}{8}$
53	$\frac{1}{4}$	$\frac{3}{8}$	$\frac{1}{8}$
54	$\frac{3}{8}$	$\frac{1}{2}$	$\frac{1}{8}$
55	$\frac{1}{2}$	$\frac{5}{8}$	$\frac{1}{8}$
56	$\frac{5}{8}$	$\frac{3}{4}$	$\frac{1}{8}$
57	$\frac{3}{4}$	$\frac{7}{8}$	$\frac{1}{8}$
58	$\frac{7}{8}$	$1$	$\frac{1}{8}$
59	$1$	$1\frac{1}{8}$	$\frac{1}{8}$
599	$1\frac{1}{8}$	$1\frac{3}{4}$	$\frac{1}{8}$

## Ford eases "ground flying" with Timken Bearings



In the landing wheels of the Ford ship, where major considerations are *anti-friction* and *endurance* under extremes of radial, thrust and resultant loads, Timken outstanding ability again is demonstrated.

Timken-equipped wheels with their extra load area and high safety factor overcomes variations in landing positions and in terrain. The exclusive combination of Timken tapered construction, Timken *POSITIVELY ALIGNED ROLLS* and Timken-made steel provides enduring economy under all operating conditions.

It is now more important than ever for aircraft engineers to have the complete Timken data before them.

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Wherever he lands he finds business.

Consider his experience at the Normal School in Fayetteville, N. C. He flies in from the North . . . a speak in the sky. Coming in, he circled the town a few times. And when he landed, a group of interested students were waiting for him.

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And almost before his motor was cold, he happened off again, circled some more and pointed his WACO South . . . and was gone. That's what you call modern selling!



The value of a WACO as an aid to Mr. Mason's sales is reflected in his record. He consistently leads the entire sales force of his company. His profits from the prestige gained through the use of his WACO . . . by his ability to cover more territory, more frequently and in less time.

A great number of business men use WACO airplanes because WACO provides fast, trustworthy, economical transportation. Hundreds of flying field operators and sportsmen use WACOs because these ships are sturdy, tough, easy to fly and can be landed in the smallest of fields. All told, more WACOs are in use today than any other make of commercial and pleasure aircraft.

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